

NATIONAL AERONAUTICS AND SPACE ADMINISTRATION

1 INTRODUCTION AND PURPOSE

FINAL EXPERIMENTS MISSION RULES

APOLLO 16

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A ABBREVIATIONS AND ACRONYMS

B DISTRIBUTION LIST

DECEMBER 15, 1971

PREPARED BY

FLIGHT CONTROL DIVISION

C CHANGE CONTROL



MANNED SPACECRAFT CENTER HOUSTON, TEXAS

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FINAL EXPERIMENTS MISSION RULES

APOLLO 16 ALSEP 3/P&FS 2

REV A

PREFACE

THIS DOCUMENT CONTAINS REVISION A TO THE EXPERIMENTS MISSION RULES FOR ALSEP 3 AND P&FS 2 AS OF MARCH 1, 1972. ALL SUBSEQUENT REVISIONS TO THIS DOCUMENT WILL BE PRINTED ON DIFFERENT COLORED PAPER FOR EASY RECOGNITION.

THIS DOCUMENT HAS BEEN PREPARED BY THE FLIGHT CONTROL DIVISION, MANNED SPACECRAFT CENTER, HOUSTON, TEXAS, WITH TECHNICAL ASSISTANCE BY SERVICE TECHNOLOGY CORPORATION.

IT IS SUGGESTED THAT ANY ORGANIZATION HAVING COMMENTS, QUESTIONS, OR SUGGESTIONS CONCERNING THESE MISSION RULES CONTACT MR. JOHN H. TEMPLE, FLIGHT CONTROL OPERATIONS BRANCH, BUILDING 30, ROOM 2064A, 713-483-3838.

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ACTING DIRECTOR OF FLIGHT OPERATIONS

SJOBERG/

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REVISION INSTRUCTION SHEET

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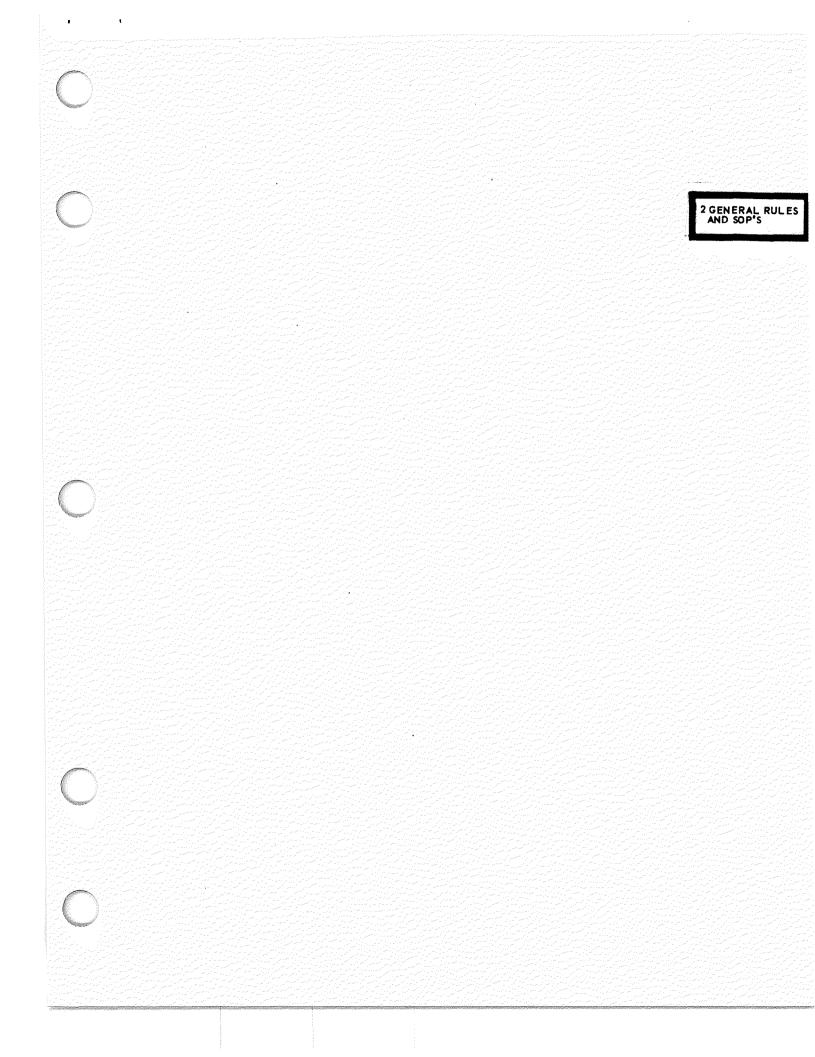
NASA -- MSC

1 INTRODUCTION AND PURPOSE

MISSION RULES

SECTION 1 - INTRODUCTION AND PURPOSE

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					INTR	ODUCTION AND PURPOSE							
													
		DECISION-MAKI AND CONSTRAIN TEXAS, HAS TH FOR ALSEP AND	NG PROCESS. TS, AND MISSI E OVERALL RES P&FS.	THE RUON OBJ	OLES ARE BA DECTIVES. BILITY FOR	SED ON AN ANALYSIS (THE DIRECTOR OF FLIT THE PREPARATION, CO	OF MISSION EQUIPM GHT OPERATIONS, M NTENTS, AND CONTR	ENT CONFIGU ANNED SPACE DL OF THE E	INES TO EXPEDITE THE RATION, SYSTEMS OPERATIONS CRAFT CENTER, HOUSTON, XPERIMENT MISSION RULES				
		DIFFERENCE IN	LIFETIME OF	THESE	EXPERIMENT		CIFIC MISSION-ORI		ON RULES BECAUSE OF THE IVITIES. THE EXPERIMENT				
						RFACE WITH THE ALSEP FACE WITH THE ALSEP		SYSTEMS LIF	FETIME				
		RULES IN THE OF CROSSREFER	FLIGHT MISSIC RENCE. THE FL	N RUL	ES AND THE MISSION RUL	EXPERIMENT MISSION	RULES DOCUMENT AR	E ASSIGNED	CREW INTERFACE. ALSEP THE SAME NUMBERS FOR EASE (IST BETWEEN THIS DOCUMENT				
						•							
		MISSION RULES CAN BE CATEGORIZED AS GENERAL AND SPECIFIC. GENERAL MISSION RULES CONTAIN THE BASIC PHILOSOPHIES USED IN THE DEVELOPMENT OF THE EXPERIMENT MISSION RULES. SPECIFIC MISSION RULES PROVIDE THE BASIC CRITERIA FROM WHICH REAL-TIME DECISIONS ARE MADE AND ARE FORMATTED AS FOLLOWS:											
		A. THE"CONDITION/MALFUNCTION" COLUMN DEFINES THE FAILURE.											
		B. THE "PHASE" COLUMN IDENTIFIES THE TIME INTERVAL IN WHICH THE CONDITION/MALFUNCTION OCCURS.											
		C. THE "RUL THE COND		FINES	FLIGHT CO	NTROLLER ACTION AND	OR PROCEDURES THA	T MUST BE	ACCOMPLISHED AS A RESULT OF				
			S/NOTES/COMMEN			IDES THE FLIGHT CONT	FROLLER WITH ADDIT	IONAL INFO	RMATION CONCERNING THE				
		THERE WILL B	E A SEPARATE	SET OF	MISSION R	ULES FOR THE EXPERIM	MENT PACKAGES FOR	EACH MISSI	on.				
-	L	<u>L</u>	MISSION	REV	DATE	SECTION	GROUP	PAGE					
			APOLLO 16	FNL	12/15/71	INTRODUCTION AND	<u> </u>	1-1					



MISSION RULES

SECTION 2 - GENERAL RULES AND SOP'S

_		. SECTION 2 - GENERAL RULES AND SOP'S
R	ITEM	
		GENERAL
		West 18 / V JL
ll	1	
	2-1	THE EXPERIMENT MISSION RULES OUTLINE PREPLANNED DECISIONS DESIGNED TO MINIMIZE THE AMOUNT OF REAL-TIME RATIONALIZATIO
		REQUIRED WHEN NON-NOMINAL SITUATIONS OCCUR DURING AND AFTER CREW ACTIVATION OF THE ALSEP OR DEPLOYMENT OF THE P&FS.
		THE PARS.
		•
	2-2	WHEN A CONFLICT OF PLANNED EXPERIMENT PACKAGE ACTIVITIES OCCUR, THE LUNAR EXPERIMENTS OFFICER WILL DETERMINE THE
		PRIORITY OF ACTIVITIES, WITH CONCURRENCE OF THE SCIENCE TEAM LEADER.
	2-3	IN SOME INSTANCES, THE SPECIFIC MISSION RULES MAY DEVIATE FROM THE GENERAL GUIDELINES OR FROM THESE GENERAL RULES.
	- "	THE SPECIFIC MISSION RULE WILL APPLY IN ALL CASES, AND THE DEVIATIONS FROM THE GENERAL GUIDELINES WILL BE NOTED.
1 1		THE STATE OF THE PARTY TO ALL CHOICES, AND THE DETERMINES FROM THE GENERAL GOIDELINES WILL BE NOTED.
	2-4	THE ALSEP SENIOR ENGINEER OR SATELLITE COMMUNICATIONS ENGINEER MAY, AFTER ANALYSIS OF THE OPERATION, CHOOSE TO TAKE
		ANY NECESSARY ACTION REQUIRED FOR SUCCESSFUL COMPLETION OF EXPERIMENT TEST OBJECTIVES.
1 1		
1 1		
	2-5	MISSION RULE LIMITS THAT ARE CONSIDERED TO BE INTERIM OR UNCONFIRMED NUMBERS WILL BE UNDERLINED IN THIS PUBLICATION
		AND ALL SUBSEQUENT REVISIONS UNTIL THE NUMBERS ARE CONFIRMED BY THE RESPONSIBLE NASA AGENCY.
		The first file of the first file of the fi
	2-6	THE SYSTEMS LIMITS LISTED IN THESE RULES ARE THE ACTUAL VEHICLE LIMITS (AS WELL AS THEY ARE KNOWN AND UNDERSTOOD)
	· 1	AND ARE NOT BIASED TO COMPENSATE FOR TIME DELAYS OR INSTRUMENTATION ERRORS WITHIN THE EXPERIMENT AND MSFN DATA/DISPLA SYSTEMS.
	1	3131 cm ₃ .
1 1		
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3 ALSEP OPERATIONAL GUIDELINES

MISSION RULES

SECTION 3 - ALSEP OPERATIONAL GUIDELINES

R ITEM												
	ALSEP OPERATIONAL GUIDELINES											
32-1	GENERAL											
	A. THESE ALSEP GENERAL OPERATIONAL GUIDELINES ARE BASED ON OBJECTIVES IN THE FOLLOWING PRIORITIES:											
	1. HFE											
	2. LSM											
	3. PSE 4. ASE											
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	NOTE RIPPLE-OFF SEQUENCE IS:											
	- Barrian de la companya de la comp - Tanggara de la companya de la comp											
	1. AFE											
	2. LSM 3. PSE											
	B. THE GATHERING OF SCIENTIFIC DATA WILL NOT BE COMPROMISED FOR ENGINEERING OR TEST PURPOSES.											
	C. REDUNDANT OR BACKUP SYSTEMS WILL NOT BE SELECTED UNLESS A FAILURE WARRANTS SUCH ACTION. SWITCHING TO REDUNDANT											
	SYSTEMS WILL NOT BE ACCOMPLISHED TO SATISFY ENGINEERING TESTS UNLESS ALL SCIENTIFIC MISSION OBJECTIVES HAVE BEEN COMPLETED.											
	D. DELETE.											
	E. BEFORE IMPLEMENTING ANY MISSION RULE ACTION BASED ON AN APPARENT ALSEP MALFUNCTION, IT WILL BE ASCERTAINED THAT THERE IS NO PROBLEM WITH THE MSFN SUPPORTING SITE.											
	F. THE TIMER INHIBIT COMMAND WILL NOT BE SENT PRIOR TO ALL EXPERIMENTS BEING ON.											
	G. THE ASE WILL BE COMMANDED TO STANDBY IF "THUMPER ARM" OR "GRENADE ARM" STATE IS INDICATED WITHOUT GROUND COMMAN OR CREW ACTION.											
	H. THE RESETTABLE TIMER WILL NORMALLY BE RESET TO ZERO DURING SUNRISE TERMINATOR SUPPORT PERIODS.											
	I. ALSEP EXPERIMENTS WILL NOT BE COMMANDED TO "STANDBY OFF" UNLESS THE ACTION IS JUSTIFIED BY AN ANOMALY.											
	J. ANYTIME THERE IS AN AUTOMATIC SWITCHOVER TO PCU 2 NOT IDENTIFIABLE TO A FAILURE IN PCU 1, A ONE-TIME COMMAND WI BE ATTEMPTED TO RETURN TO PCU 1 IF THE +12 VDC BUS IS GREATER THAN 11.8 VDC.											
	K. NO COMMAND FUNCTION CAN BE EXECUTED (OTHER THAN "STANDBY OFF," "STANDBY SELECT," OR "OPERATE SELECT") IN AN EXPERIMENT, BY GROUND COMMAND OR BY ONBOARD TIMER, UNLESS THE EXPERIMENT IS IN THE "OPERATE" MODE.											
	L. THE ALSEP SHORTING PLUG SWITCH AND THE ASTRONAUT SWITCH 1 WILL BE ACTIVATED IN THIS ORDER ASAP AFTER DEPLOYMEN											
	M. IF THE GROUND IS UNABLE TO COMMAND TRANSMITTER A "ON" AND/OR EXPERIMENTS "ON," THE GROUND WILL REQUEST THE ASTRONAUT TO TURN ON ASTRONAUT SWITCHES 2 AND/OR 3.											
	N. THE CENTRAL STATION HEATERS AND PDR WILL BE USED TO MAINTAIN AN AVERAGE INTERNAL THERMAL PLATE TEMPERATURE GREATER THAN 20° F AND LESS THAN 125° F UNLESS THERE IS AN ANOMALY REQUIRING THE CS HEATERS TO BE OFF AND/OR THE POWER IS REQUIRED TO MAINTAIN EXPERIMENT INTEGRITY.											
<u> </u>	MISSION REV DATE SECTION GROUP PAGE											
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MISSION RULES

R ITEM	SECTION 3 - ALSEP OPERATIONAL GUIDELINES - CONTINUED
32-1 CONT	O. A SINGLE COMMAND CARRIER WILL BE MAINTAINED "ON" AND IN THE SAFED CONFIGURATION, EXCEPT TO SEND COMMANDS AND REMOTE SITE HANDOVERS FROM DEPLOYMENT INITIATION UNTIL AFTER LM LIFT-OFF IN ORDER TO PROVIDE ADDITIONAL PROTECTION AGAINST THE GENERATION OF SPURIOUS COMMANDS.
	P. THE HFE BORE HOLES WILL HAVE PRIORITY OVER THE DRILL CORE STEM. THE HFE BORE HOLES WILL BE ATTEMPTED FIRST. PROBLEMS ARE ENCOUNTERED, EFFORTS ON BORE HOLES WILL BE TERMINATED AFTER A TOTAL OF 20 MINUTES ACCUMULATED DRILL ON TIME.
	Q. IF A HARD OBJECT IS ENCOUNTERED THAT REDUCES DRILL RATE TO LESS THAN APPROXIMATELY 5 INCHES PER MINUTE ON EITH HFE PROBE HOLE, THE FOLLOWING WILL BE ACCOMPLISHED:
	 IF THE SECOND STEM SECTION IS NOT ATTACHED, WITHDRAW AND START AT A DIFFERENT LOCATION FOR MAXIMUM OF TWO WITHDRAWALS.
	2. IF THE SECOND STEM IS ATTACHED, CONTINUE UNTIL APPROXIMATELY 10 MINUTES OF POWER ON TIME FOR THE DRILL STRING HAS ELAPSED.
	R. HFE CONDUCTIVITY MEASUREMENTS IN MODE II AND MODE III WILL BE COMPLETED PRIOR TO TEMINATION OF CONTINUOUS REAL-TIME SUPPORT.
	S. IF THE CREW MUST RETURN TO THE LM PRIOR TO COMPLETE ALSEP DEPLOYMENT, THE SHORTING PLUG SWITCH AND THE ASTRONAUT SWITCH 1 WILL BE ACTIVATED "ON" IF THE ANTENNA IS EMPLACED. IF THE ANTENNA IS NOT EMPLACED, THESE SWITCHES WILL NOT BE ACTIVATED (PICK UP HERE ON EVA 2).
	T. WHENEVER CMD 003 (HBR ON) IS USED TO GO TO HIGH BIT RATE, CMD 005 (HBR OFF) WILL BE USED TO RETURN TO NORMAL BIT RATE (TO PREVENT LOSS OF CVW'S)
	U. THE ASE IS THE ONLY EXPERIMENT THAT WILL BE COMMANDED WHILE IN HIGH BIT RATE.
	V. THE PSE WILL NOT BE ACTIVATED BY GROUND COMMAND UNTIL THE ASE THUMPER ACTIVITY HAS BEEN COMPLETED.
	W. THE ALSEP WILL BE IN HIGH BIT RATE DURING LM LIFT-OFF (-5 MINUTES TO NOMINAL +15 MINUTES).
	X. FOR ANY MALFUNCTION DURING A SURFACE TASK, A MAXIMUM OF 10 MINUTES WILL BE SPENT ON THE CONTINGENCY PROCEDURE BEFORE THE TASK IS ABANDONED, WITH THE FOLLOWING EXCEPTIONS:
	1. RTG FUELING - UP TO 20 MINUTES WILL BE ALLOWED IN EXERCISING RTG FUELING CONTINGENCY PROCEDURES.
	 ALSEP PACKAGE 1 TO PACKAGE 2 CABLE CONNECTIONS - UP TO 20 MINUTES WILL BE ALLOWED FOR MAKING THE CABLE CONNECTION.
	3. ALSEP ANTENNA - UP TO 30 MINUTES WILL BE ALLOWED FOR ANTENNA ERECTION AND ALIGNMENT.
	4. ALSEP DEPLOYMENT MAY BE CONTINUED ON A LATER EVA IF ADDITIONAL TIME TO SPEND ON CONTINGENCY PROCEDURES IS REQUIRED TO ATTAIN AN OPERATIONAL ALSEP.
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MISSION RULES

SECTION 3 - ALSEP OPERATIONAL GUIDELINES - CONTINUED

32-1 CONT	Υ.	FOR EVA TERMINATION OR OTHER INTERRUPTIONS DURING ALSEP DEPLOYMENT, THE FOLLOWING DEPLOYMENT INTERRUPTION POINTS WILL BE OBSERVED IF PERMITTED BY CREW SAFETY CONSIDERATION:									
		 REMOVE ALSEP PACKAGES 1 AND 2. CLOSE SEQ BAY DOOR. REPLACE ALSEP PACKAGES WITH HANDLES UP AND WITH EXPERIMENTS FACING THE SUN WITHIN ±15°. 									
		2. TILT FUEL CASK (DOME NOT REMOVED).									
		3. TILT FUEL CASK. REMOVE DOME. DO NOT DEFUEL.									
		4. FUEL RTG. THEN CARRY ALSEP TO DEPLOYMENT SITE. REMOVE ALL COMPONENTS FROM PACKAGE 2. CARRY PACKAGE 1 TO EMPLACEMENT SITE. DO NOT CONNECT RTG CABLE TO CS UNLESS SUFFICIENT TIME IS AVAILABLE TO ERECT THE CS SUNSHIELD.									
		5. CONNECT HFE CABLE TO CS. REMOVE PSE, ASE, AND LSM FROM SUBPACKAGE 1. ALIGN CS AND RAISE SUNSHIELD. MOUNT ANTENNA MAST, GIMBAL, AND ANTENNA. LEVEL AND ALIGN ANTENNA. DEPRESS SHORTING PLUG AND ROTATE ASTRONAUT SWITCH 1 CLOCKWISE ON WAY BACK TO LM.									
		6. THE HFE CAN BE INTERRUPTED AFTER COMPLETION OF THE FIRST PROBE HOLE.									
		7. DEPLOY ALSEP EXPERIMENTS AND COMPLETE TASKS. A HOLD POINT EXISTS AFTER EACH TASK IS COMPLETED. DEPRESS SHORTING PLUG SWITCH AND ROTATE ASTRONAUT SWITCH 1 CLOCKWISE ON WAY BACK TO LM. (SEE CONTINGENCY PROCEDURES FOR HOLD POINTS.)									
	z.	IF ALL DOTS ON TEMPERATURE LABLE ON HORSE COLLAR ARE BLACK AFTER FUELING RTG, CREW MUST NOT TOUCH SHORTING PLUG UNTIL IT HAS COOLED SUFFICIENTLY (10 MINUTES ON PACKAGE 1).									
	AA. ALSEP DEPLOYMENT WILL NOT BE STARTED IF IT IS KNOWN THAT LESS THAN 1 HOUR 30 MINUTES REMAIN IN										
	AB.	TRAVERSES IN FRONT (±30° OF THE MORTAR FIRE LINE) OF THE GLA WILL BE ALLOWED IF GLA ENABLE AND ACTIVATION SWITCHES ARE DISABLED AND ASTRONAUT SWITCH 5 IS CLOCKWISE (SAFED).									
32-2	PSE										
	A.	THE PSE WILL BE UNCAGED ASAP AFTER THE PSE IS COMMANDED TO "OPERATE SELECT".									
	В.	THE HEATERS WILL NOT BE TURNED OFF PRIOR TO LEVELING OPERATIONS, WITH THE FOLLOWING EXCEPTIONS:									
		1. DURING INITIAL LEVELING OPERATIONS 2. WHEN PSE IS CURRENT LIMITED									
	c.	THE PSE LEVELING MOTORS WILL NOT BE ACTIVATED PRIOR TO UNCAGING.									
	D.	AFTER PSE LEVELING OF THE HORIZONTAL COMPONENTS (LPX AND LPY) HAS BEEN COMPLETED, THE COARSE LEVEL SENSORS WILL BE GROUND COMMANDED OUT (VIA CMD 102).									
	Ε.	FOR PROPER OPERATION OF THE PSE, THE FEEDBACK FILTER MUST BE OUT DURING LEVELING OPERATIONS.									
	F.	FOR INITIAL LEVELING, THE X-AXIS AND Y-AXIS MASSES OF THE PSE SHOULD BE LEVEL BEFORE LEVELING THE Z-AXIS MASS.									
	G	THE PSE WILL BE RELEVELED AS INFREQUENTLY AS POSSIBLE.									
		합니다. 그 나 아들들이 맛있는데 나는 사람들은 사람들은 사람들은 사람들은 사람들은 사람들은 사람들은 사람들은									

MISSION RULES

SECTION 3 - ALSEP OPERATIONAL GUIDELINES - CONTINUED

ITEM	SECTION 3 - ALSEP OPERATIONAL GUIDELINES - CONTINUED
32-2 CONT	H. DURING FORCED LEVELING OPERATIONS, CAUTION SHALL BE EXERCISED PRIOR TO INITIATING LEVELING MOTOR OPERATION TO INSURE THAT PROPER MODE, DIRECTION, AND SPEED HAVE BEEN SELECTED.
	I. THE PSE WILL BE COMMANDED TO STANDBY PRIOR TO ASE MORTAR FIRINGS. THE COMMAND WILL BE SENT PRIOR TO HIGH BIT RATE.
32-3	ASE
	A. OPERATION OF THE ASE, IN ANY MODE, WILL BE PERFORMED ONLY WHEN AS-1 (CENTRAL STATION PACKAGE TEMPERATURE) IS ABOVE -20° C, AS-2 (MORTAR BOX TEMPERATURE) IS ABOVE -60° C, AND AS-3 (GRENADE LAUNCHER ASSEMBLY TEMPERATURE) IS ABOVE -60° C BEFORE MORTAR FIRING.
	B. OPERATION OF THE ASE IN THE MORTAR MODE WILL BE PERFORMED ONLY WHEN AS-2 (MORTAR BOX TEMPERATURE) AND AS-3 (GRENADE LAUNCHER ASSEMBLY TEMPERATURE) ARE ABOVE -20° C AND BELOW +85° C AT START OF LAUNCHING.
	C. FOR OPERATION OF THE ASE IN THE THUMPER MODE, ADEQUATE TIME WILL BE ALLOWED AFTER ASE ACTIVATION TO PERMIT GEOPHONE AMPLIFIER TEMPERATURES TO STABILIZE. (SEE SODB, VOL. IV, P. E-4-11.)
	D. A MINIMUM WAITING PERIOD OF 60 SECONDS WILL BE ALLOWED BETWEEN ARMING AND FIRING OF THE MORTARS (FOR CHARGING OF CAPACITORS).
	E. THE INDIVIDUAL FIRE COMMANDS WILL BE THE NOMINAL MODE OF FIRING THE ASE MORTARS.
	F. ONCE MORTAR FIRING HAS COMMENCED, ALL MORTARS WILL BE FIRED WITHIN THAT LUNATION (BECAUSE OF THE LOSS OF THERMAL INTEGRITY CAUSED BY THE FIRING OF THE FIRST MORTAR).
	G. THE ASE MORTARS WILL BE FIRED ON THE THIRD LUNAR DAY FOLLOWING DEPLOYMENT, AFTER THE FOLLOWING CONDITIONS HAVE BEEN MET:
	1. TWO 85-FOOT SITES ARE SCHEDULED FOR NOT LESS THAN 3 HOURS OF SIMULTANEOUS ALSEP SUPPORT.
	2. THE NORMALIZED SUN ANGLES ARE BETWEEN 15° AND 75°.
	3. THE GLA AND MORTAR BOX TEMPERATURES ARE BETWEEN +85° AND -20° C.
	H. IF THE MORTAR PACKAGE IS DEPLOYED OFF NOMINAL, THEN GRENADE FIRING WILL BE DELAYED UNTIL THE PROBABLE EFFECTS OF FIRING HAVE BEEN STUDIED.
	I. THE ASE WILL BE TURNED TO "STANDBY OFF" EXCEPT DURING LISTENING MODES AFTER ASE MORTARS HAVE BEEN FIRED.
	J. THE ASE GRENADE LAUNCHING WILL BE CONSIDERED PRIOR TO THE SCHEDULED TIME IF THE ALSEP IS IN DANGER OF IMMINENT TOTAL OR PARTIAL FAILURE. THE CONDITIONS OF IMMINENT TOTAL FAILURE ARE:
	1. LOSS OF ONE TRANSMITTER
	2. LOSS OF ONE PCU 3. EXTENDED, INCREASING, ERRATIC PERFORANCE OF UPLINK
	4. LOSS OF DATA PROCESSOR
	5. LOW RESERVE POWER INSUFFICIENT TO MAINTAIN PSE OPERATION, ASE STANDBY MODE, AND HFE OR LSM.
	THE CONDITION OF PARTIAL FAILURE IS COMPLETE FAILURE OF ANY TWO OF THE OTHER THREE EXPERIMENTS.
	MISSION REV DATE SECTION GROUP PAGE APOLLO 16 A 3/1/72 ALSEP OPS GENERAL 3-4
	APOLLO 10 A 3/1/72 ALSEP OF 3 GENERAL 3-4 Tape

MISSION RULES

R ITEM											
32-3 CONT		THE ASE WILL BE OF REMAIN ABOVE -60°				INIMUM OF 30 MINUI	ES EACH WEEK,	PROVIDED AS 02 AND 03			
			POLLO PROGR					LLO SPACECRAFT PROGRAM , THEN THE FOLLOWING			
		1. LUNAR SCIENCE	ACTIVITY !	IS NEAR A !	MINIMUM.						
		2. TWO 85-FOOT S	ITES ARE SO	CHEDULED F	OR NOT LESS THAN	3 HOURS OF SIMULTA	NEOUS ALSEP SI	UPPORT.			
		3. THE ALSEP EXP	ERIMENTS AI	RE NOT PIC	KING UP TERMINATO	R CROSSING ACTIVIT	Y (WITHIN 2 D/	AYS OF TERMINATOR).			
		4. THE GLA AND MO	ORTAR BOX	TEMPERATUR	ES ARE BETWEEN 15	.5° AND 26.6° C.					
		5. ALL MORTARS W	ILL BE FIRI	ED AND IN	THE FOLLOWING ORD	ER: -2, -4, -1, -	3.				
				SEQU IS -	NOTE ENTIAL FIRE ORDER 2, -4, -3, -1.						
	ASE GRENADE LAUNCHING APPROVAL WILL BE REQUESTED IF THE ALSEP IS IN DANGER OF IMMINENT TOTAL OR PARTIAL FAILURE. THE CONDITIONS OF IMMINENT TOTAL FAILURE ARE: 1. LOSS OF ONE TRANSMITTER AND ERRATIC PERFORMANCE IN THE SECOND 2. LOSS OF ONE PCU AND ERRATIC PERFORMANCE OF SECOND PCU 3. EXTENDED, INCREASING, ERRATIC PERFORMANCE OF UPLINK 4. LOSS OF DATA PROCESSOR AND POOR PERFORMANCE OF SECOND UNIT 5. LOW RESERVE POWER INSUFFICIENT TO MAINTAIN PSE OPERATION, ASE STANDBY MODE, AND ANY OF THE FOLLOWING: SIDE/CCIG OR CPLEE										
		THE CONDITIONS OF	PARTIAL F	AILURE ARE							
		1. COMPLETE FAIL 2. COMPLETE FAIL									
		THE ASE MORTARS W OPERATING PSE EXP		LAUNCHED	IF THE APOLLO 14	PSE EXPERIMENT EX	ISTŞ AS ONE OF	TWO REMAINING PAIRS OF			
	М.	IF THE GROUND IS TO TURN ON ASTRON			and the second second second second	l "HBR", THEN THE C	GROUND WILL RE	QUEST THE ASTRONAUT			
	N.	IF THE CREW DOES BE REDUCED TO ALL				CACTIVITY AND ARM	THE GLA, THUM	MPER ACTIVITY TIME WILL			
	0.	IF AN ASI FAILS T WALK TO NEXT FIRI				BE MADE TO FIRE TH	E ASI. IF A S	SECOND NO-FIRE OCCURS,			
	Р.	IF INSUFFICIENT T	IME IS AVA	ILABLE FOI	R COMPLETE THUMPE	R ACTIVITY, THE FO	LLOWING POSITI	IONS WILL BE USED:			
		1. FIVE MINUTES 2. TEN MINUTES -			, 16, 18, 21						
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MISSION RULES

SECTION 3 - ALSEP OPERATIONAL GUIDELINES - CONCLUDED

NO FLIP/CAL WILL SE ALLOWED SETWERN THE X-, Y-, AND Z-SITE SURVEYS. THRRE-INITIATED FLIP/CALS SHALL BE INHIBITED (BY LSN FLIP/CAL INHIBIT) UNTIL COMPLETION OF THE SITE SURVEY. B. OPERATIONAL POWER WILL NOT BE REMOVED FROM THE LSN DURING THE LUNAR WIGHT PERIODS, EXCEPT AS NOTED IN RULE 32-62. C. SITE SURVEY WILL NOT BE PLANNED UNTIL AFTER LM ASCENT AND FOUR FLIP/CALS HAVE BEEN SENT. D. ONE FLIP/CAL WILL BE PLANNED PRIOR TO LM ASCENT. E. THE LSM WILL BE ON DURING LPM OPERATIONS. 32-5 HFE A. SEQUENTIAL COMMAND UPLINIZED TO THE HEE WILL BE SEPARATED BY AT LEAST 54 SECONDS IN NORMAL BIT RATE AND 108 SECONDS IN LOW BIT RATE. B. A CONDUCTIVITY MEASUREMENT WILL NOT BE INITIATED UNLESS THERE WILL BE SUFFICIENT POWER TO COMPLETE THE MEASUREMENT WITHOUT INTERQUIPTION. ONCE A PROBE HEATER IS TUNNED ON FOR AN EXPERIMENT, IT WILL NOT BE TUNNED OF UNLESS THE COMMOUGHTUITY MEASUREMENT IS TO BE TERMINATED, OR OTHER ALSEP CONTINGENCIES ARE TO BE CORRECTED. C. THE HEE WILL BE COMMANDED TO STANDBY PRIOR TO ASE MORTAR FIRING. 32-6 INSUFFICIENT POWER FOR SIMULTANEOUS SUPPORT OF ALL EXPERIMENTS AUTO THERMAL CONTROL OF THE CENTRAL STATION AND AUTO THERMAL CONTROL OF THE EXPERIMENTS WILL BE INHIBITED IF ADEQUATE POWER IS NOT AVAILABLE. THERMAL CONTROL AND THE MANUALLY MANUACED TO PRECLUDE REPORTED FOR OPERAL STATION AURE AS PERMEMBERS WILL BE ALLOWED TO GO AS LOW AS -20° F IF THE HEATER POWER IS REQUIRED FOR OPERAL STATION AURE AS PERMEMBERS WILL BE ALLOWED TO GO AS LOW AS -20° F IF THE HEATER POWER IS REQUIRED FOR OPERAL OF THE INTERFERENCE THAN ANOTHER EXPERIMENT OF ANY LUNAR DAY (29.5 EARTH DAYS). RULE NUMBERS 32-B THROUGH 32-10 ARE RESERVED. NISSION REV DATE. SECTION GROUP PAGE	TEM											
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4 ALSEP SPECIFIC RULES

MISSION RULES

SECTION 4 - ALSEP SPECIFIC RULES

R	RULE	CONDITION/MALFUNCTION	PHASE		RULING	CUES/	NOTES/COMMENTS
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		RULE NUMBERS 32-12 THROUGH 32-20 ARE RESERVED.					
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MISSION RULES

SECTION 4 - ALSEP SPECIFIC RULES - CONTINUED

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								AE-7	+29	+31.3	+25.7	
1								AE-8	+15	+16.2	+13.8	
1 1					1			AE-10	+5	+ 5.4	+ 4.6	
								AE-11	-12	-11.0		
					- 1			AE-12		- 5.5		
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MISSION RULES

SECTION 4 - ALSEP SPECIFIC RULES - CONTINUED .

R	RULE	CONDITION/MALFUNCTION	PHAS	E	RULING	CUES/NOTES/COMMENTS
	32-31	ALSEP FAILS TO RESPOND TO A COMMAND		Α.	REINITIATE THE COMM/	NO FUNCTIONAL VERIFICATION AND NO COMMAND VERIFICATION WORD (CVW)
				В.	IF UNSUCCESSFUL, SEE OTHER DECODER AND RE THE COMMAND	ECT
	32-32	FAILURE OF 18-HR TIMER		A.1	. INITIAL CONTINUOUS	A.1. THE TIMER WILL BE DEEMED FAILED AFTER
					SUPPORT: IF ANY TWO OF THE ! ING TEMPS ARE BETWE AND +155° F, CONTIN REAL-TIME SUPPORT 1 45 DAYS:	EN -20° IUE
			:		AT-31, CMD DEC BASE AT-32, CMD DEC INT AT-33, CMD DEC VCO	T
				2	AFTER THE INITIAL CONTINUOUS SUPPORT XMTR WILL BE LEFT (PROVIDING THE ALSEF	DN.
					(A) RETURNING VALID SCIENTIFIC DATA	
					(B) THERE IS NO IND OF FAILURE OR I FAILURE IN THE SYSTEM	MINENT CONSIDERATION IN PREDICTING THE
					STOTEM	1. ARE THE COMMAND SYSTEM TEMPS (AT-31, AT-32, AT-33) CONSISTENT WITHIN ±5° F OF THEIR VALUES AT THE SAME SUN ANGLE DURING THE PREVIOUS LUNAR DAY.
				ĺ		2. IS THE ALSEP RESPONDING NORMALLY TO ALL COMMANDS.
						3. IS A CVW BEING RECEIVED AFTER A MINIMUM OF 90 PERCENT OF THE COMMANDS TRANSMITTED.
						CONTINUES THANSPITTED.
		RULE NUMBERS 32-33 AND 32-34 ARE RESERVED.				
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		MISSIO	N REV	DATE	SECTION	GROUP PAGE
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R	RULE	CONDITION/MALFUNCTION	PHASE		RULING		CUES/NOTES/COMMENTS
	32-35	WEAK TM SIGNAL		A.	SELECT REDUNDANT XI	ITR	A. XMTR A SEL - CMD 012 XMTR B SEL - CMD 015
				В.	SELECT LOW BIT RATE		B. LOW BIT RATE SEL - CMD 007
	32-36	LOSS OF SYNC OR BAD DECOMMED DATA		A.	SELECT REDUNDANT PP	ROCESSOR	A. PROC X SEL - CMD 034 PROC Y SEL - CMD 035
				В.	SELECT LOW BIT RATE		AB-6 - D/P X ON/OFF STATUS B. LOW BIT RATE SEL - CMD 007
A	32-37	LOSS OF TM SIGNAL		지원 보관	IF IN HBR, SELECT I	NBR	A. HBR OFF - CMD 005
Α					SEND TM ON SELECT REDUNDANT PO	CU .	B. TM ON - CMD 013 C. PCU 2 SEL - CMD 062 (GROUP 2) PCU 1 SEL - CMD 060
				D.	SELECT REDUNDANT XI	4TR	D. XMTR A SEL - CMD 012 XMTR B SEL - CMD 015 NOTE
A							IF PSE LEVELING IS IN PROGRESS, SEND PSE STBY SEL - CMD 037. IF HFE IS IN MODE II OR III, CMD MODE I (CMD 135).
	32-38	DATA DEMAND SIGNAL FROM DATA PROCESSOR FAILS HIGH		USI	ECT REDUNDANT PROCES NG THE OTHER DECODE RETURN TO FAILED PR	R. DO	CUE CVW IS STEADILY INDICATING AN ERRONEOUS PATTERN OTHER THAN ALL ZEROS (CVW LIGHT ON EVERY 0.6 SEC) AFTER A COMMAND HAS BEEN SENT. DISABLE COMMAND TO FAILED PROCESSOR. PROC X SEL - CMD 034 PROC Y SEL - CMD 035
A	32-39	CRAIND IMARIE TO					
	32-39	GROUND UNABLE TO COMMAND HIGH BIT RATE OFF		A.	SEE RULE 32-31		
Α					CMD 3A-005 (HBR OFI		
Α				D.	CMD 38-011 (NBR RES		
				E.	CMD 3A-035 (OR 3A-034)		DSS/PROC Y SEL - CMD 035
				F.	CMD 3B-035		DSS/PROC X SEL - CMD 034
				6	(OR 3B-034) IF CREW IS AVAILABLE	F HAVE	G. SW NO. 5 WILL BE TURNED CCW PRIOR TO
		RULE NUMBERS 32-40 THROUGH 32-45 ARE			SW NO. 5 TURNED CW		CREW LEAVING THE ALSEP AREA FOR THE FINAL
		RESERVED.					
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MISSION RULES

ì	RULE	CONDITION/MALFUNCTI	ON PI	HASE		RULING		CUES/NOTES/COMMENTS
	32-46	FAILURE OF AUTOMATIC LEVELING MODE				ORM FORCED LEVELING COMMANDS	3 вү	PSE ACTIVATION PRESETS LEVELING MODE TO AUTOMATIC. CMD 103 WILL SWITCH TO FORCED MODE. CMD 103 IS A TWO-STATE COMMAND - PSE
	32-47	PSE LEVELING MOTOR FAILS ON		٠		ND COMMAND PSE TO CT AND THEN TO OPE CT		CUE LEVELING MODE AUTO/FORCED. CUE LEVELING MOTOR WILL DRAW 3 W FROM RESERVE POWER. PSE STBY SEL - CMD 037
	32-48	PSE LEVELING MOTOR FAILS OFF			NO AC	CTION TO BE TAKEN		PSE OPER SEL - CMD 036 CUE NO DROP IN RESERVE POWER (NORMALLY 3 W) WHEN REPEATED EFFORTS ARE MADE TO TURN MOTOR ON, AND NO ACTIVITY ON SHORT-PERIOD Z-DATA CHANNEL.
	32-49	FAILURE OF MECHANICA LEVEL DRIVE	L		AND	CT LOW SPEED AND H DIRECTION REVERSAL: RNATELY		CUE NO MOTION OF THE PLATFORM CAN BE DETECTED ON THE LP HORIZONTAL OR LP-Z COMPONENT TIDAL OUTPUT. X-MTR ON/OFF - CMD 070
								Y-MTR ON/OFF - CMD 071 Z-MTR ON/OFF - CMD 072 DIRECTION PLUS/MINUS - CMD 074 SPEED LOW/HIGH - CMD 075
	32-50	MISALIGNED COARSE SENSOR			B. F	COMMAND COARSE SEN: WHEN CORRESPONDING IS IN LEVELING OPEI PERFORM FORCED LEVI OF THE PSE BY GROUI COMMAND	MOTOR CATION ELING	WHILE IN FINAL LEVELING PHASE (LOW STEPPING RATE), THE MOTOR REVERTS TO A HIGH STEPPING RATE REPEATEDLY WITHOUT ACHIEVING CENTERING. COARSE LEVEL SENSOR AND GIMBAL WILL NEVER ALIGN, AND THE MOTOR WILL CONTINUE TO DRIVE BEYOND LEVEL. COARSE SENSOR OUT - CMD 102.
	32-51	FAILURE OF COARSE LE	VEL		B. 6	SELECT FORCED PSE IMODE GROUND COMMAND COAISENSOR OUT 1. PROCEED WITH II FORCED LEVELING COARSE LEVELING	RSE LEVEL NITIAL G FOR	A. PSE LEVELING MODE AUTO/FORCED - CMD 103 B. COARSE LEVEL SENSOR OUT - CMD 102 CUE NO HIGH SPEED MOTOR OPERATION IS NOTICED DURING INITIAL LEVELING PHASE, AND COMPONENT DOES NOT CENTER WITHIN EXPECTED TIME (35 MIN MAXIMUM IN AUTO MODE). USE HIGH SPEED LEVELING FOR COARSE LEVELING.
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MISSION RULES

R	RULE	CONDITION/MALFUNCTION	PHASE			RULING		CUES/NOTES/COMMENTS
	32-52	LONG-PERIOD COMPONENT	,		Δ !	SE HIGH SPEED, FO	PCFD	
		STICKS			B. I	SE HIGH SPEED, TO EVELING IN DIRECT ULLS MASS AWAY FF F UNSUCCESSFUL, S OW SPEED AND HIGH ND DIRECTION REVE LITERNATELY	ION THAT OM STOP ELECT SPEED	CUE FAILURE TO CENTER WITHIN EXPECTED TIME (35 MIN MAXIMUM IN AUTO MODE). NOTE DO NOT EXCEED 5 MIN 30 SEC IN HIGH SPEED. X-MTR ON/OFF - CMD 070 Y-MTR ON/OFF - CMD 071 Z-MTR ON/OFF - CMD 072 DIRECTION PLUS/MINUS - CMD 074 SPEED HIGH/LOW - CMD 075
	32-53	ELECTRICAL FAILURE OF LONG-PERIOD COMPONENT	:		AFFEC	NATE LEVELING OF TED AXIS AFTER CO OR PHASE IS COMPLE	ARSE	CUE TIDAL OUTPUT IS WITHIN RANGE, BUT IS NOT AFFECTED BY LEVELING OR CENTERING DRIVE.
	32-54	AUTOMATIC SWITCHOVER OF PSE TO STANDBY			COMMA	IND PSE TO OPERATE	SELECT	CHECK RESERVE POWER. IF CIRCUIT BREAKER CB-06 HAS OPENED FROM OVERCURRENT (500 MA ± 10%), STANDBY MODE WILL BE SELECTED, AND THE CB WILL BE RESET AUTOMATICALLY. PSE OPER SEL - CMD 036
	32-55	FAILURE OF PSE UNCAGE SEQUENCE			B. 1	O UNCAGE ARM: SEND UNCAGE AR IF UNSUCCESSFI 18-HR TIMER PI ARM ACTUATOR IF UNSUCCESSFI 144 HR +2 MIN FROM DELAYED (SEQUENCER WILLI ACTUATOR O UNCAGE FIRE (AR CTUATOR HAS BEEN SEND UNCAGE AR FINENCESSFI PSE STEY SEL THEN PSE OPER (CMD 036) IF UNSUCCESSFI 18-HR TIMER PI FIRE THE ACTUAL	IL, FIRST ILSE WILL IL, PULSE OMMAND ARM TER ARMED): IM/FIRE IL, SEND CMD 037), SEL IL, NEXT ILSE MAY	NORMAL UNCAGING IS ACCOMPLISHED BY SENDING CMD 073, UNCAGE ARM/FIRE, TWICE: ONCE TO ARM AND ONCE TO FIRE THE ACTUATOR. NOTE UNCAGING MAY NOT BE POSSIBLE UNLESS AT-05 THERMAL PLATE 3 TEMP IS ABOVE +25° F. NOTE SELECTING PSE STANDBY MAY FIRE ACTUATOR IF IT HAS BEEN ARMED AND NOT FIRED. NOTE 30 SEC IS REQUIRED BETWEEN ARM AND FIRE TO CHARGE CAPACITOR.
		MISSION	REV	DATE		SECTION	GROUP	PAGE
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MISSION RULES

	C RULES - CONTINUED

R	RULE	CONDITION/MALFUNCTION	PHASI		RULING		CUES/NOTES/COMMENTS
	32-56	PSE GOES OFF WHILE IN STANDBY MODE		DI	JRING NORMAL OPERATION DMMAND PSE TO STANDBY	S, SELECT	CUE EXPERIMENT 1 STANDBY DISCRETE EXTINGUISHED, AND RESERVE POWER INCREASED SINCE POWER IS REMOVED FROM THE HEATERS. IF FUSE (F-03) HAS BEEN BLOWN BY OVERCURRENT (500 MA), CAPABILITY TO SELECT PSE STANDBY MODE IS LOST. PSE STBY SEL - CMD 037
	32-57	PSE TEMPERATURE LOW AND AUTO THERMAL CONTROL FAILS		CC	OMMAND HEATER TO FORCE	D ON	ASSUME AUTOMATIC THERMOSTAT CONTROL FAILED. CMD 076 IS A FOUR-STATE COMMAND, WHICH CAN SEQUENTIALLY STEP THROUGH THE FOLLOWING MODES TO CONTROL THE PSE SENSOR HEATER: 1. AUTO OFF 2. FORCED HTR ON 3. FORCED OFF 4. AUTO ON
A							DL-07 PSE INSTRUMENT TEMP LOW LIMIT IS +125° F. MINIMUM OF 8 W OF RESERVE POWER IS REQUIRED.
	32-58	PSE TEMPERATURE HIGH AND AUTO THERMAL CONTROL FAILS.		A	COMMAND HEATER TO FI	DRCED	A. CMD 076 IS A FOUR-STATE COMMAND: 1. AUTO OFF 2. FORCED HTR ON 3. FORCED OFF 4. AUTO ON DL-07 PSE INSTRUMENT TEMP HIGH LIMIT IS +127° F.
				В	IF UNSUCCESSFUL, COI PSE TO STANDBY, THEI OPERATE STANDBY	MMAND N TO	B. SELECTING PSE TO STANDBY WILL REINITIATE TO AUTOMATIC THERMOSTAT CONTROL.
	32-59	LOSS OF DOWNLINK DURING LEVELING MOTOR OPERATION RULE NUMBER 32-60 IS RESERVED.		SI	END PSE STBY SEL		NOTE PSE STBY SEL WILL STOP MOTORS.
		MISSION APOLLO 16	REV	DATE 3/1/72	SECTION 2 ALSEP SPECIFIC RULES	GROUP PSE	PAGE 4-7 Tape 2-5

MISSION RULES

32-62 FAIL STOF	NOT MAINTAIN LSM SOR TEMPERATURE AT 40° O° C P/CAL SEQUENCE NOT TIATED BY TIMER AT HR INTERVALS	SELECT	COMMAND THE FOLLOWING SEQUENCE A. SEND FLIP/CAL INHIBIT B. LSM STBY (CMD 046) A. VERIFY FLIP/CAL INHIBIT OFF B. COMMAND FLIP/CAL GO C. IF UNSUCCESSFUL, SEND FOLLOWING COMMAND SEQUENCE 1. STBY SEL (CMD 046)	NORMAL FLIP/CAL SEQUENCE IS 350 SEC. DURING LUNAR NIGHT, RETURN TO OPERATE WITHIN 1 MIN AFTER STANDBY. SEND STANDBY COMMAND ONLY AT THE END OF A FLIP/CAL SEQUENCE. CUE DM-23 IS A ZERO - CMD 127 B. CMD 131 C. REINITIALIZE LOGIC
32-63 FLIF	P P/CAL SEQUENCE NOT TIATED BY TIMER AT		A. SEND FLIP/CAL INHIBIT B. LSM STBY (CMD 046) A. VERIFY FLIP/CAL INHIBIT OFF B. COMMAND FLIP/CAL GO C. IF UNSUCCESSFUL, SEND FOLLOWING COMMAND SEQUENCE	NORMAL FLIP/CAL SEQUENCE IS 350 SEC. DURING LUNAR NIGHT, RETURN TO OPERATE WITHIN 1 MIN AFTER STANDBY. SEND STANDBY COMMAND ONLY AT THE END OF A FLIP/CAL SEQUENCE. CUE DM-23 IS A ZERO - CMD 127 B. CMD 131 C. REINITIALIZE LOGIC
INIT	TIATED BY TIMER AT		OFF B. COMMAND FLIP/CAL GO C. IF UNSUCCESSFUL, SEND FOLLOWING COMMAND SEQUENCE	DM-23 IS A ZERO - CMD 127 B. CMD 131 C. REINITIALIZE LOGIC
			2. OPER ON (CMD 045) 3. FLIP/CAL INHIBIT OFF (CMD 127) 4. FLIP/CAL GO (CMD 131)	
32-64 FAIL MAIN	LURE OF HEADS TO NTAIN SAME DIRECTION		SEND UP TO FIVE FLIP/CAL'S IN AN ATTEMPT TO ALIGN HEADS	CUES: DM - 9 DM - 10 DM - 11
THRO	E NUMBERS 32-65 OUGH 32-70 ARE ERVED.			
		REV DAT	TE SECTION GROU 1/72 ALSEP SPECIFIC LSM	IP PAGE 4-8

MISSION RULES

R	RULE	CONDITION/MALFUNCTION	PHASE		RULING		CUES/NOTES/COMMENTS
	32-71	IT IS NECESSARY TO DISARM A GRENADE			SMIT CMD 043 TO TUR TANDBY	N ASE	LEAVE IN STANDBY FOR AT LEAST 3 MIN.
	32-72	GRENADE FIRE COMMAND FAILS TO FIRE A GRENADE		1	SEE RULE 32-31 USE CMD 162 (ASE SE	Q FIRE)	B. WHEN USING CMD 162 TO FIRE GRENADE, THE SEQUENCE IS 2, 4, 3, 1.
	32-73	DURING DEPLOYMENT, GROUND IS UNABLE TO COMMAND TO HIGH BIT RATE (HBR)		İ	SEE RULE 32-31 HAVE THE CREW ACT SW NO. 4 AND COMP THUMPER ACTIVITY		SW NO. 4 WILL TURN THE ASE TO ON AND SELECT HBR.
			•	2.	AT COMPLETION OF ACTIVITY, COMMAND OFF (CMD 005) PRI CREW TURNING SW N CW. COMPLETE NOR DEPLOPYMENT.	HBR OR TO O. 5	
				3.	DURING FINAL EVA, CREW TURN SW NO. AND ACTIVATE SW N	5 CCW	
				4.	CONDUCT MORTAR EX AFTER LIFT-OFF AN TO LM ASCENT STAG	D PRIOR	
				5.	GROUND COMMAND TO LM ASCENT STAGE I		B.5. THERE ARE THREE COMMAND ROUTES TO RETURN TO NBR: 005, 011, AND 034/035.
	32-74	DURING DEPLOYMENT, GROUND IS UNABLE TO		Α.	SEE RULE 32-31		
		COMMAND FROM HIGH BIT RATE TO NORMAL BIT RATE		B.1.	IF DURING THUMPER ACTIVITIES, FIRST COMPLETE THE THUM MODE IN HBR		B.1. SUCCESSFUL THUMPING REQUIRES HBR.
				2.	AT COMPLETION OF THUMPING MODE, HA STANDBY PRIOR TO SW NO. 5 CW WHILE ALTERNATE COMMAND CHECKED	TURNING	
				3.	COMMAND NBR RESET		
				4.	IF 3 IS UNSUCCESS COMMAND PROCESSOR (CMD 035)		
				5.	IF EITHER 3 OR 4 SUCCESSFUL, CONTI NORMAL MISSION		B.5. DISABLE CMD 003 IN THE R/S RTC INVENTORY.
				6.	IF 4 IS UNSUCCESS HAVE CREW TURN SW CW AND CONTINUE N EVA	NO. 5	B.6. GRENADE FIRING MODE WILL BE CONDUCTED PER MISSION RULE GUIDELINE 32-3, EXCE FIRING WILL BE CONDUCTED USING PSE AS THE SENSOR IN NBR.
				7.	DO NOT COMMAND HB	R ON	
		MISSION	REV	DATE	SECTION	GROUP	PAGE
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MISSION RULES

R	RULE	CONDITION/MALFUNCTION	PHASE		RULING		CUES/	NOTES/COMME	NTS	
	32-75	DURING DEPLOYMENT, GROUND IS UNABLE TO COMMAND ASE TO OPERATE SELECT		B. P. 1	EER RULE 32-31 PRIOR TO THUMPER MO . HAVE CREW ACTIV. SW NO. 4 AND CO WITH GEOPHONE DEPLOYMENT. PRETURN TO HER ACTIVITY. IF COMPLETE THUMPE ACTIVITY. IF UNSUCCESSFUL, S 32-73 AND 33-74 COMPLETE THUMPE ACTIVITY. COMPLETE THUMPE ACTIVITY ACTIVITY HAVE CREW TURN CW AS IN NOMINA DEPLOYMENT SEQUE AFTER GLA ARMIN SAFING PIN REMO TURN SW NO. 5 C ACTIVATE SW NO. COMMAND TO NBR	ATE NTINUE EMPT BR (CMD AND THEN CMD 003 - UCCESSFUL, R EE RULE (CMD AT HUMPER SW NO. 5 L ENCE G AND VAL, CW AND 4.		SEL, IT IS PRIOR TO CCC WHETHER COM NBR SELECT	ESSENTIAL T MPLETION OF MAND CAPABI EXISTS.	D TO ASE OPER O ESTABLISH, THUMPER ACTIVIT LITY FOR HBR AND
	32-76	GROUND UNABLE TO		CONTI	005 - HBR OFF)			SELECT, ALS SURVIVE LUN	EP IN NBR. IAR NIGHT IN ITAR FIRING	E IN OPERAIE (ASE WILL NOT OPERATE SELECT. MODE PRIOR TO
		COMMAND ASE TO STANDBY		OPERA	ATE MODE					
A	32-77	ASI FAILS TO FIRE		AT TH SECON NEXT	MPT ONCE TO FIRE AS HE SAME LOCATION. HD NO-FIRE OCCURS, FIRING LOCATION AN MEXT ASE.	IF A WALK TO				
A	32-78	AFTER THE FIRING OF ONLY ONE OR ANY COMBINATION OF MORTARS, ASE PITCH (DS-7) OR ASE ROLL (DS-6) HAVE CHANGED BY ±5°			ONTINUE MORTAR LAUN ENCE UNTIL DATA CAN JATED					
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MISSION RULES

R	RULE	CONDITION/MALFUNCTION	PHASE	, gain 12.	RULING	CUES/NOTES/COMMENTS	
	32-79	PRIOR TO ANY MORTAR FIRING, ASE ROLL (DS-6) AND/OR ASE PITCH (DS-7) PARAMETERS INDICATE SENSOR FAILURE			FIRE FIRST MORTAR, DISF ING SENSOR INDICATION CONTINUE WITH REMAINING MORTARS AFTER ASSURING RT DATA THAT OTHER INDI TIONS ARE NOMINAL	FROM	
STATES OF STREET, SAME OF STREET, OF STREET, OF STREET, OF STREET, OF STREET, OF STREET, OF STREET, OF STREET,	32-80	UNEXPLAINED LOSS OF LOCK ON HIGH BIT RATE			COMMAND ASE TO STANDBY SELECT (CMD 043) THEN TO OPERATE SELECT (CMD 042 SELECT ALTERNATE PROCES 1. PROC X SEL (CMD 034 2. PROC Y SEL (CMD 035	O PRESENT. COMMANDING A RESET CIRCUIT BREAKERS TO THE ASE 16-CHANNEL SOR: B. DO NOT APPLY IF MISSIO BEEN INVOKED.	SE TO STANDBY WILL THAT PROVIDE POWER MULTIPLEXER.
		RULE NUMBERS 32-81 THROUGH 32-100 ARE RESERVED.					
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		MISSION	REV	DATE	SECTION	ROUP	

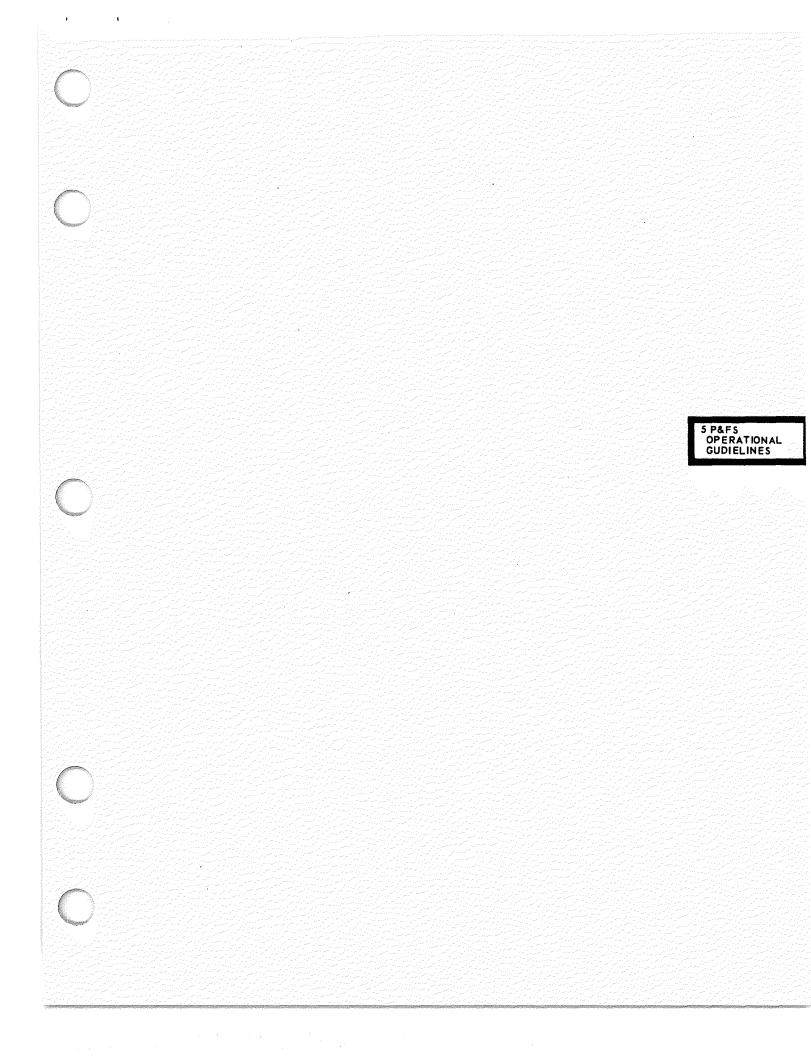
MISSION RULES

R RULE	CONDITION/MALFUNCTION	PHASE	RULING	CUES/NOTES/COMMENTS
32-10	UNABLE TO DRILL NORMAL HFE EMPLACEMENT HOLES			
	A. IF THE HOLE IS LESS THAN 24 IN. DEEP		A. DIG TRENCH APPROXIMATE 4 FT LONG, SLOPING FROI 1 IN. DEEP AT ONE END APPROXIMATELY 18 IN. D. AT THE OTHER END. PLA PROBE IN TRENCH WITH C. AT SHALLOW END. COVER PROBE AND FIRST 6 FT OI CABLE WITH AS MUCH SOIL POSSIBLE.	1 TO TO EEP CE ABLE
A	B. IF THE HOLE IS NOT NORMAL DEPTH		B. PLACE PROBE IN HOLE AS AS IT WILL GO. EMPLAC LOWER RADIATION SHIELD THE TOP OF THE PROBE R GARDLESS OF PROBE DEPTI	ON _
32-10	HAVE CHOICE OF DRILLING 2ND HFE HOLE OR CORE SAMPLE HOLE		DRILL 2ND HFE PROBE EMPLAC HOLE	EMENT HEE HAS PRIORITY OVER CORE SAMPLE.
32-10:	DRILL RATE REDUCED TO LESS THAN 5 IN./MIN		A. IF LESS THAN TWO STEM SECTIONS ARE ATTACHED THE POWER HEAD, WITHDR AND START AT NEW LOCAT FOR MAXIMUM OF TWO WITHDRAWALS	
A			B. IF TWO OR MORE STEM SECTIONS ARE ATTACHED THE POWER HEAD, CONTINI UNTIL 10 MIN OF POWER- TIME FOR THE DRILL STR HAS ELAPSED	JE DN
32-104	HFE INTERRUPTED DURING CONDUCTIVITY MEASUREMENT (HEATER ON) A. DURING MODE II:			
	1. HEATER GOES OFF		A.1. GO TO NEXT CONDUCTIV MEASUREMENT	ITY
	2. HEATER GOES TO MODE III		2. TURN HEATER OFF AND RETURN TO MODE II AN TO NEXT HEE CONDUCTI' MEASUREMENT	
A			IF ON TIME IS MORE TO 6 HR, GO TO DECAY MO IF ON TIME IS LESS TO 6 HR, GO TO NEXT CONDUCTIVITY MEASURED	DE. HAN
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MISSION RULES

SECTION	4 - ALSEP	SPECIFIC	RULES -	CONCLUDED
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R	RULE	CONDITION/MALFUNCTION	PHASE		P SPECIFIC RULES -		CUES/NOTES/COMME	NTS
	32-105	HFE DOWNLINK DATA LOSES SYNC		В	IF RULE 32-38 HAS NO INVOKED AND CONDUCTI MEASUREMENT IS IN PROMITCH DATA PROCESSO 1. DATA PROC X SEL (CMD 034) 2. DATA PROC Y SEL (CMD 035) IF BETWEEN HEATER ON SEND HFE STBY, THEN COMMANDS	VITY OGRESS, R:	B. HFE STBY - CM HFE ON - CMD	
A	32-106	HFE ELECTRONICS REFERENCE TEMPERATURE GREATER THAN 333° K		GO TO	D STBY OFF FOR 12 HR		CUES: DH-13 T ₁ REF DH-15 T ₂ REF HFE STBY - CMD 05: HFE ON - CMD 052	3
		RULE NUMBERS 32-107 THROUGH 32-110 ARE RESERVED.						
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					RULES			Tape 3-1



NASA - Manned Spacecraft Center MISSION RULES

SECTION 5 P&FS OPERATIONAL GUIDELINES

R ITEM	SECTION 5 P&FS OPERATIONAL GUIDELINES								
	P&FS OPERATIONAL GUIDELINES								
	FOR P&FS MANAGEMENT RULES AND GUIDELINES PRIOR TO DEPLOYMENT, REFERENCE THE OSO - SPECTROMETERS GO CHART IN SECTION 3 OF THE APOLLO 16 FMR DOCUMENT.								
	RULE NUMBERS 33-1 THROUGH 33-8 ARE RESERVED.								
33-9	WATER AND URINE DUMPS WILL BE INHIBITED FROM 1 HOUR BEFORE UNTIL IMMEDIATELY AFTER THE P&FS LAUNCH.								
	RULE NUMBERS 33-10 THROUGH 33-12 ARE RESERVED.								
33-13	ONE EVA FOR FILM RETRIEVAL WILL BE CONSIDERED IN THE EVENT THE SUBSATELLITE FAILS TO COMPLETE A SUCCESSFUL LAUNCH SEQUENCE.								
	RULE NUMBERS 33-14 THROUGH 33-17 ARE RESERVED.								
33-18	FOR A NONLANDING MISSION, THE SUBSATELLITE DESIRED TRAJECTORIES ARE LISTED BELOW IN THEIR ORDER OF SCIENCE PRIORITY:								
	A. ELLIPTICAL (LUNAR) B. ELLIPTICAL (EARTH) C. SOLAR ORBIT								
33-19	THE P&FS OPERATIONAL GUIDELINES ARE BASED ON THE FOLLOWING OBJECTIVES, WHICH ARE LISTED IN THEIR ORDER OF PRIORITY:								
	A. PARTICLES EXPERIMENT S-173 B. FIELD EXPERIMENT S-174 C. S-BAND TRACK S-164								
33-20	HARDWARE REDLINES WILL NOT BE VIOLATED IN THE PURSUIT OF SCIENCE DATA UNLESS THE DATA BEING COLLECTED ARE JUDGED TO BE MORE IMPORTANT THAN ALL SUBSEQUENT DATA.								
	4 - 아프로그램, 이 사람들은 그는 그렇게 한국을 받는 이번에 하는 것으로 하는 아픈 아픈 아픈 아픈 아들이 사람들이 되었다. 그는 그렇게 하는 아픈 아들이 아들이 아들이 아들이 아들이 아들이 아들이 아들이 아들이 아들이								
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SECTION 5 - P&FS OPERATIONAL GUIDELINES - CONTINUED

R	ITEM			,						
	33-21	THE GATHERING BY THE WORK SCI	OF SCIENTIFIC HEDULE.	DATA	WILL NOT B	E COMPROMISED FOR E	NGINEERING/TEST PU	RPOSE UNLE	SS SPECIFICALLY	/ DIRECTED
	33-22	NO MISSION RUL EXISTS WITH SU	E BASED UPON PPORTING GROU	A P&FS ND EQI	S MALFUNCTI JIPMENT.	ON WILL BE IMPLEMEN'	TED UNTIL IT HAS BE		AINED THAT NO F	PROBLEM
							- 70 1			
							•			
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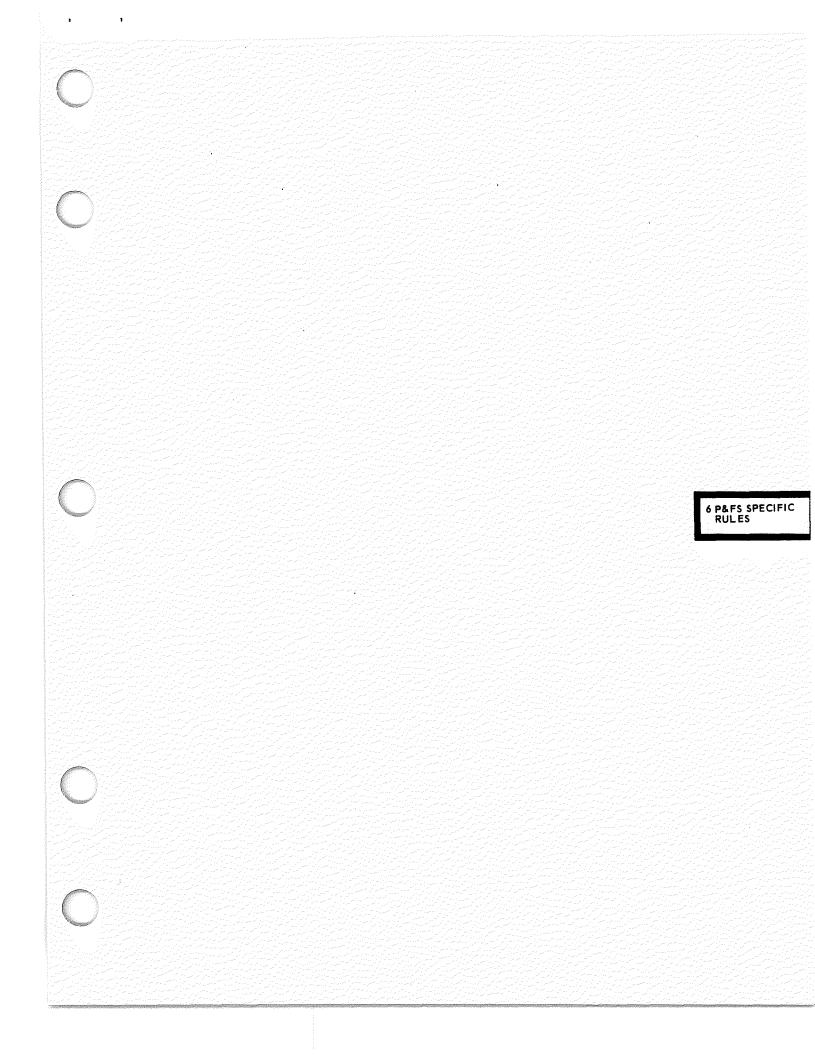
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MISSION RULES SECTION 5 - P&FS OPERATIONAL GUIDELINES - CONTINUED ITEM P&FS MANAGEMENT THE INITIAL P&FS ACTIVATION WILL BE PERFORMED IN SUNLIGHT. 32-23 A 33-24 THE P&FS HIGH VOLTAGE WILL NOT BE ACTIVATED UNTIL P&FS LAUNCH + 48 HOURS. THE EXPERIMENT/DATA OFF COMMAND WILL NOT BE TRANSMITTED UNLESS THE ACTION IS REQUIRED TO ENHANCE P&FS 33-25 OPERATIONS OR IS JUSTIFIED BY AN ANOMALY AFFECTING THE ACCOMPLISHMENT OF THE MISSION. 33-26 THE P&FS BATTERY WILL NOT NORMALLY BE DEPLETED BELOW 5.0 AMPERE-HOURS, BUT A DEPLETION OF 75 PERCENT APPROXIMATELY ONCE PER MONTH WILL BE ALLOWABLE. THE UNDERVOLTAGE PROTECTION CIRCUIT (UVP) WILL BE DISABLED IF THE BATTERY VOLTAGE IS LESS THAN 11 VDC AND THE 33-27 REMAINING LIFE OF THE SATELLITE IS PREDICATABLY LIMITED OR BELIEVED TO BE IN JEOPARDY, OR IF A UVP MALFUNCTION IS SUSPECTED TO HAVE OCCURRED. 33-28 IN NO INSTANCE WILL P&FS OPERATION BE PERMITTED TO INTERFERE WITH THE LM OPERATION. THE SATELLITE WILL BE COMMANDED OFF PRIOR TO APOLLO LAUNCHES. IF THE COMMAND FUNCTION FAILS, THE AUTO KILL CIRCUIT 33-29 WILL BE ALLOWED TO DEACTIVATE THE SATELLITE. RULE NUMBERS 33-30 THROUGH 33-94 ARE RESERVED. 33-95 THE P&FS WILL BE LAUNCHED NORTHWARD PERPENDICULAR TO THE ECLIPTIC PLANE. THE P&FS WILL BE LAUNCHED INTO AN ORBIT WITH A REVOLUTION PERIOD OF 7134 ± 30 SECONDS. 33-96 SUBSATELLITE LAUNCH WILL NOT BE ATTEMPTED WITH THE GRS BOOM EXTENDED. 33-97

MISSION	REV	DATE	SECTION	GROUP	PAGE	
 APOLLO 16	Α	3/1/72	P&FS OPS	GENERAL	5-3	
	報節	Capatrian A	GUIDELINES		9-46	Tape 3-4

NASA - Manned Spacecraft Center MISSION RULES

		SEC	TION 5 -	P&FS OP	ERATIONAL GUIDELINES	S - CONCLUDED		
R	ITEM		er de la ferma		and the second s	566s	traci	ida
	33-98	SUBSATELLITE LAUNCH WILL NO	T BE CONS	TRAINED	BY THE FAILURE OF	EXPERIMENT COVERS.		
Α	33-99	EFFLUENT DUMPS WILL BE AVOI THE START OF THE P&FS LAUNC	DED FOR 1 H SEQUENC	HOUR P	RIOR TO P&FS LAUNCH. AFTER P&FS LAUNCH.	AND ALL RCS JETS	WILL BE IN	HIBITED (CMC FREE) FROM
	33-100	IMMEDIATELY BEFORE AND AFTE BELOW 5.0 AMPERE-HOURS, THE	R ENTERIN BATTERY	G INTO WILL BE	A FLIGHT MODE THAT I	WILL RESULT IN A D Y CHARGED STATE.	EPLETION O	F THE BATTERY TO
		RULE NUMBERS 33-101 THROUGH	33-199 A	IRE RESE	RVED.			
1								
1		MISSION		ATE	SECTION	GROUP	PAGE	and the state of the same of t
		APOLLO 16	A 3/	1/72	P&FS OPS GUIDELINES	GENERAL	5-4	Tape 3-5



MISSION RULES

SECTION 6 - P&FS SPECIFIC RULES

R	RULE	CONDITION/MALFUNCTION	PHASE	}	RULING		CUES/NOTES/COMMENTS
	33-200	TRANSPONDER ON LOGIC FAILURE	LO	CON	TINUE MISSION		TRACKING LIMITED TO RT AND MRO OPERATIONAL MODES
	33-201	REDUCED TRANSMITTER OUTPUT A. IF UPLINK ON B. IF UPLINK OFF	L0	\ ^.	TURN OFF UPLINK TURN ON UPLINK		A. SELECTS TX 95.1 MHZ L.O. B. SELECTS RX 95.1 MHZ L.O. REF MALF PROCEDURE
	33-202	DOWNLINK CARRIER FREQUENCY DEVIATION A. IF UPLINK ON B. IF UPLINK OFF	L0	1	TURN OFF UPLINK TURN ON UPLINK		A. SELECTS TX 95.1 MHZ L.O. B. SELECTS RX 95.1 MHZ L.O. REF MALF PROCEDURE
	33-203	LOSS OF DOWNLINK CARRIER A. IF UPLINK ON B. IF UPLINK OFF	LO		TURN OFF UPLINK TURN ON UPLINK		A. SELECTS TX 95.1 MHZ L.0 B. SELECTS RX 95.1 MHZ L.0. REF MALF PROCEDURE
	33-204	LOSS OF MESSAGE ACCEPTANCE PULSE	LO	CON	TINUE MISSION, USIN RRIDE FUNCTION	G MAP	SATELLITE LIFE MAY BE LIMITED DUE TO POSSIB POSSIBLE FAILURE TO RESET AUTO KILL CIRCUIT
		RULE NUMBERS 33-205 THROUGH 33-210 ARE RESERVED.					
		WISSION	REV	DATE	SECTION	GPOUP	PAGE
		APOLLO 16	FNL	12/15/71	P&FS SPECIFIC RULES	COMM/INS	

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MISSION RULES

SECTION 6 - P&FS SPECIFIC RULES - CONTINUED .

33-213 BATTERY TEMPERATURE LO MANAGE POWER TO MINIMIZE PROBLEM 33-214 EXCESSIVE BATTERY CURRENT DUE TO TRANSMITTER MALFUNCTION 33-215 EXCESSIVE DURRENT DUE TO DITHER THAN THE ABOVE (MILES 33-212 AND 23-213) 33-216 EXCESSIVE URRENT DUE TO DITHER THAN THE ABOVE (MILES 33-212 AND 23-213) 33-217 BATTERY VOLTAGE BELOW 11.2 V WHILE DISCHARGE (MILES 33-212 AND 23-213) 33-216 LOSS OF OUTPUT FROM A SOLAR CELL STRING 33-217 BATTERY SHUNT FALLS LO CONTINUE MISSION WITH POSSIBLE RESCHEDULING 33-218 BATTERY SHUNT FALLS LO CONTINUE MISSION WITH POSSIBLE BY 2.5 PERCENT 33-217 BATTERY SHUNT FALLS LO CONTINUE MISSION WITH POSSIBLE BY 2.5 PERCENT 33-218 BATTERY CHARGE CONTROL LO CONTINUE MISSION WITH POSSIBLE BY 2.5 PERCENT 33-218 BATTERY CHARGE CONTROL LO CONTINUE MISSION WITH POSSIBLE BY 2.5 PERCENT 33-218 BATTERY CHARGE CONTROL LO CONTINUE MISSION 33-218 BATTERY CHARGE CONTROL LO CONTINUE MISSION 33-219 BATTERY CHARGE CONTROL LO CONTINUE MISSION 33-210 BATTERY CHARGE CONTROL LO CONTINUE MISSION 33-210 BATTERY CHARGE CONTROL LO CONTINUE MISSION 33-210 BATTERY CHARGE CONTROL LO CONTINUE MISSION 33-210 BATTERY CHARGE CONTROL LO CONTINUE MISSION 33-210 BATTERY CHARGE CONTROL LO CONTROL MILL PROTECT BATTERY. 33-210 BATTERY CHARGE CONTROL LO CONTROL MILL PROTECT BATTERY. 33-210 BATTERY CHARGE CONTROL LO CONTROL MILL PROTECT BATTERY. 33-210 BATTERY CHARGE CONTROL LO CONTROL MILL PROTECT BATTERY. 33-210 BATTERY CHARGE CONTROL LO CONTROL MILL PROTECT BATTERY. 33-210 BATTERY CHARGE CONTROL LO CONTROL MILL PROTECT BATTERY. 33-2110 BATTERY CHARGE CONTROL LO CONTROL MILL PROTECT BATTERY. 33-212 BATTERY CHARGE CONTROL LO CONTROL MILL PROTECT BATTERY. 33-213 BATTERY CHARGE CONTROL LO CONTROL MILL PROTECT BATTERY. 33-214 BATTERY CHARGE CONTROL LO CONTROL MILL PROTECT BATTERY. 33-215 BATTERY CHARGE CONTROL LO CONTROL MILL PROTECT BATTERY. 33-216 BATTERY CHARGE CONTROL LO CONTROL MILL PROTECT BATTERY. 33-217 BATTERY SHUNT FALLS 33-218 BATTERY CHARGE CONTROL LO CONTROL MILL PROTECT BATTERY. 33-219 BATTERY CHAR	ITS	NOTES/COMMENTS	CUES/NC		RULING		Ξ.	PHASE	FUNCTION	CONDITION/MAL	RULE	R
CURRENT DUE TO TRANS- MITTER MALFUNCTION 33-213 EXCESSIVE BATTERY CURRENT DUE TO ELECTRO- STATIC ANALYZERS (EA) H/ MALFUNCTION 33-214 EXCESSIVE CURRENT DUE TO OTHER THAN THE ABOVE (RULES 33-212 AND 33-213) BATTERY VOLTAGE BELOW 11.2 V WHILE DIS- CHARGING CONTINUE MISSION WITH POSSIBLE RESCHEDULING CONTINUE MISSION WITH POSSIBLE RESCHEDULING ACONTINUE MISSION WITH POSSIBLE RESCHEDULING CONTINUE MISSION WITH POSSIBLE BY 2.5 PERCENT BATTERY SHUNT FAILS OPEN CONTINUE MISSION WITH POSSIBLE BY 2.5 PERCENT BATTERY SHUNT FAILS OPEN CONTINUE MISSION BATTERY CELL OVERCHARGE PROTECTION C CONTINUE MISSION BATTERY CELL OVERCHARGE PROTECTION C CONTINUE MISSION BATTERY CELL OVERCHARGE PROTECTION C CHARGE VOLTAGE CHARGE CHARGE CHARGE CHARCH VOLTAGE CHARGE CHARCH CHARCH CHA				NIMIZE			-	LO	ATURE	BATTERY TEMPER >90° OR <10° F	33-211	
33-213 EXCESSIVE BATTERY CLAURENT DUE TO ELECTRO-STATIC ANALYZERS (EA) HV MALFUNCTION 33-214 EXCESSIVE CURRENT DUE TO OTHER THAN THE ABOVE (RULES 33-212 AND 33-213) 33-215 BATTERY VOLTAGE BELOW 1.2 V WHILE DIS-CHARGING 33-216 LOSS OF OUTPUT FROM A SOLAR CELL STRING 33-216 ACTION ACTIO					CE USE OF TX	REDU		L0	TRANS-	CURRENT DUE TO MITTER MALFUNC	33-212	
TO OTHER THAN THE ABOVE (RULES 33-212 AND 33-212 AND 33-213) RESCHEDULING RESCHEDULING RESCHEDULING RESCHEDULING RESCHEDULING RESCHEDULING MAY EFFECT MISSION CAPABILITIES, DEPI UPON DEGREE OF FAILURE MAY EFFECT MISSION CAPABILITIES, DEPI UPON DEGREE OF FAILURE POWER AVAILABLE FROM SOLAR CELLS RED BY 2.5 PERCENT RESCHEDULING RESCHEDULING RESCHEDULING A. CONTINUE MISSION WITH POSSIBLE BATTERY CHARGE CONTROL (BCC) REGULATOR SWITCH LOGIC FAILURE A. FAILS IN 15.5 V REGULATOR POSITION RESCHEDULING RESCHEDULING A. CONTINUE MISSION WITH CHARGE CONTROL CHARGE PROTECTION CHARGE VOLTAGE. A. CONTINUE MISSION WITH CHARGE COLTAGE. B. FAILS IN 16.9 V B. CONTINUE MISSION B. BATTERY WILL BE REDUCED DUE CHARGE VOLTAGE. B. BATTERY WILL TEND TO OVERCHARGE.					OFF HVPS	TURN		LO	ELECTRO-	EXCESSIVE BATT. CURRENT DUE TO STATIC ANALYZE	33-213	
11.2 V WHILE DIS- CHARGING RESCHEDULING RESCHEDULING RESCHEDULING CONTINUE MISSION WITH POSSIBLE BY 2.5 PERCENT RESCHEDULING POWER AVAILABLE FROM SOLAR CELLS RED BY 2.5 PERCENT CONTINUE MISSION BATTERY SHUNT FAILS OPEN CONTINUE MISSION BATTERY CELL OVERCHARGE PROTECTION C WILL PROTECT BATTERY. A. CONTINUE MISSION WITH RESCHEDULING A. CONTINUE MISSION WITH RESCHEDULING A. CONTINUE MISSION WITH RESCHEDULING A. CHARGE RATE WILL BE REDUCED DUE CHARGE VOLTAGE. B. BATTERY WILL TEND TO OVERCHARGE. B. CONTINUE MISSION B. BATTERY WILL TEND TO OVERCHARGE.				ITH POSSIBLE				LO	THE ABOVE	TO OTHER THAN (RULES 33-212)	33-214	
SOLAR CELL STRING RESCHEDULING RESCHEDULING RESCHEDULING BY 2.5 PERCENT B	N CAPABILITIES, DEPENDING ILURE	FECT MISSION CAPA EGREE OF FAILURE	MAY EFFI UPON DE	ITH POSSIBLE				LO		11.2 V WHILE D	33-215	
OPEN 33-218 BATTERY CHARGE CONTROL (BCC) REGULATOR SWITCH LOGIC FAILURE A. FAILS IN 15.5 V REGULATOR POSITION B. FAILS IN 16.9 V B. CONTINUE MISSION WITH RESCHEDULING B. FAILS IN 16.9 V B. CONTINUE MISSION B. BATTERY CHARGE PROTECTION C WILL PROTECT BATTERY. A. CHARGE RATE WILL BE REDUCED DUE CHARGE VOLTAGE. B. B. CONTINUE MISSION B. BATTERY WILL TEND TO OVERCHARGE.	ROM SOLAR CELLS REDUCED	AVAILABLE FROM SO PERCENT	POWER A' BY 2.5 I	ITH POSSIBLE	INUE MISSION WI HEDULING	CONT RESC		LO	FROM A	LOSS OF OUTPUT SOLAR CELL STR	33-216	
(BCC) REGULATOR SWITCH LOGIC FAILURE A. FAILS IN 15.5 V REGULATOR POSITION B. FAILS IN 16.9 V B. CONTINUE MISSION WITH RESCHEDULING A. CHARGE RATE WILL BE REDUCED DUE CHARGE VOLTAGE. B. BATTERY WILL TEND TO OVERCHARGE,	CHARGE PROTECTION CIRCUIT ERY.	Y CELL OVERCHARGE ROTECT BATTERY.	BATTERY WILL PRO		INUE MISSION	CONT		LO	FAILS		33-217	
	E. TEND TO OVERCHARGE, BUT I	ARGE VOLTAGE. TTERY WILL TEND TO	CHAI B. BAT		RESCHEDULING			LO	PR SWITCH 15.5 V POSITION 6.9 V	(BCC) REGULATOR LOGIC FAILURE A. FAILS IN 19 REGULATOR B. FAILS IN 10	33-218	
MISSION REV DATE SECTION GROUP PAGE APOLLO 16 FNL 12/15/71 P&FS SPECIFIC EPS 6-2							-					_ _

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MISSION RULES

SECTION 6 - P&FS SPECIFIC RULES - CONTINUED

33		BCC 15.5 V REGULATOR FAILS TO REGULATE BCC 16.9 V REGULATOR FAILS TO REGULATOR FAILS TO REGULATE INVALID DETECTION OF UNDERVOLTAGE CONDITION	LO LO		RULING UE MISSION UE MISSION		BATTERY VOLTAGE WILL BE REGULATED TO HIGHER VOLTAGE LEVEL, DEPENDING UPON LOAD. BATTERY VOLTAGE WILL BE REGULATED TO HIGHER WILL BE REGULATED
33	3-220	BCC 16.9 V REGULATOR FAILS TO REGULATE INVALID DETECTION OF	LO .				VOLTAGE LEVEL, DEPENDING UPON LOAD. BATTERY VOLTAGE WILL BE REGULATED TO HIT HER
		FAILS TO REGULATE INVALID DETECTION OF		CONTIN	UE MISSION		BATTERY VOLTAGE WILL BE REGULATED TO HICHER
33	3-221	INVALID DETECTION OF	LO	- 1			VOLTAGE LEVEL, DEPENDING UPON LOAD.
				CONTIN	UE MISSION		REESTABLISH NORMAL OPERATION, DEPENDING UPOFREQUENCY OF OCCURRENCE.
ļ		RULE NUMBERS 33-222 THROUGH 33-225 ARE RESERVED.					
		MISSION	REV DA		SECTION	GROUP	PAGE.
		APOLLO 16	FNL 12		&FS SPECIFIC TULES	EPS	6-3 Tape

MISSION RULES

SECTION 6 - P&FS SPECIFIC RULES - CONTINUED

		CONDITION/MALF		PHASE			RULING						
	33-226	ANALOG-TO-DIGIT		LO		CONT	INUE MISSION				TA WILL	REQUIRE CORRECTI	ON.
		REFERENCE ERROR											
	33-227	FAILURE OF SUN	SENSOR	LO		CONT RESC	INUE MISSION I HEDULING	WITH PO	OSSIBLE	SECTORING WILL BE I	GATES A	AND B-FIELD CALCU	LATI'NS
		•		·									
	33-228	FAILURE OF BIT INDICATOR	RATE	LO		CONT	INUE MISSION			TSN (8 BP SPECIFIED	S) OR TS TO THE	SF (16 BPS) MUST ALCS.	BE MANUALLY
	33-229	FAILURE OF FORM	MAT ID	LO		CONT	INUE MISSION			RT OR MRO TO THE AL	FORMAT	MUST BE MANUALLY	SPECIFIED
										TO THE ME			
	33-230	DATA STORAGE UN FAILURE	IIT	LO		CONT	INUE MISSION			LOSS OF T IN RT MOD	S CAPABI E ONLY.	LITY. DATA MAY	BE OBTAINED
	33-231	LOSS OF TS MODE		LO			INUE MISSION,			THIS MAY	PRECLUDE	E USE OF AUTO MOD	E IF TSN
						FAIL	ED TS MODE AND	JKI MO	JUE	MODE FAIL	ED.		
	33-232	LOSS OF RT MODE		L0		CONT AND	INUE MISSION, AUTO MODES	USING	TS, MRO,	SOME LOSS SITE LOCK	OF MRO -UP DELA	DATA WILL OCCUR	DUE TO MSFN
	33-233	FAILURE TO ACHI	EVE	LO		CONT	INUE MISSION N	WITH PO)SSIBLE				
	1	12 ± 2 RPM SPIN	I RATE			RESCI	HE DUL ING						
		RULE NUMBERS 33 THROUGH 33-240 RESERVED.	3-234 ARE										
		Т										T	
-			MISSION APOLLO 16	REV	DATE	- /71	SECTION DECLETO		GROUP	P	AGE		
1		Amilian	WLOFFO 10	FNL	12/15	7//1	P&FS SPECIFIC RULES		DATA		6-4		Tape 3-9

MISSION RULES

SECTION 6 - P&FS SPECIFIC RULES - CONTINUED

R	RULE	CONDITION/MALFUNCTION	PHASE	6 - P&FS SPECIFIC RULES - CONTINU RULING	CUES/NOTES/COMMENTS
7				NOLTHO	COLS/ NOTES/ COMMENTS
	33-241	CORONA A. EA	L0	A. ATTEMPT TO REDUCE BY OUTGASSING	
				B. IF A IS UNSUCCESSFUL: 1. RESCHEDULE TO OPTIMIZE DATA BEING OBTAINED 2. TURN OFF HVPS IF NO VALID SCIENTIFIC DATA	
		B. TRANSPONDER		IS BEING OBTAINED A. ATTEMPT TO REDUCE BY OUTGASSING	
				B. IF A IS UNSUCCESSFUL, RESCHEDULE TO OPTIMIZE DATA BEING OBTAINED	
	33-242	EA HV INCORRECT	LO	CONTINUE MISSION	ENERGY RANGE OF EA'S WILL CHANGE.
	33-243	EXCESSIVE EA VETO ACTION	LO	CONTINUE MISSION	INCREASE FREQUENCY OF CALIBRATION.
	33-244	FAILURE TO TELESCOPE A/B SEL LOGIC	LO	CONTINUE MISSION	LOSS OF DATA FROM UNSELECTED TELESCOPES
	·	RULE NUMBERS 33-245 THROUGH 33-250 ARE RESERVED.			
			<u> </u>		
		MISSION APOLLO 16	REV DATE	SECTION GPOUP 5/71 P&FS SPECIFIC PARTICL	PAGE E 6-5
				RULES DETECTO	

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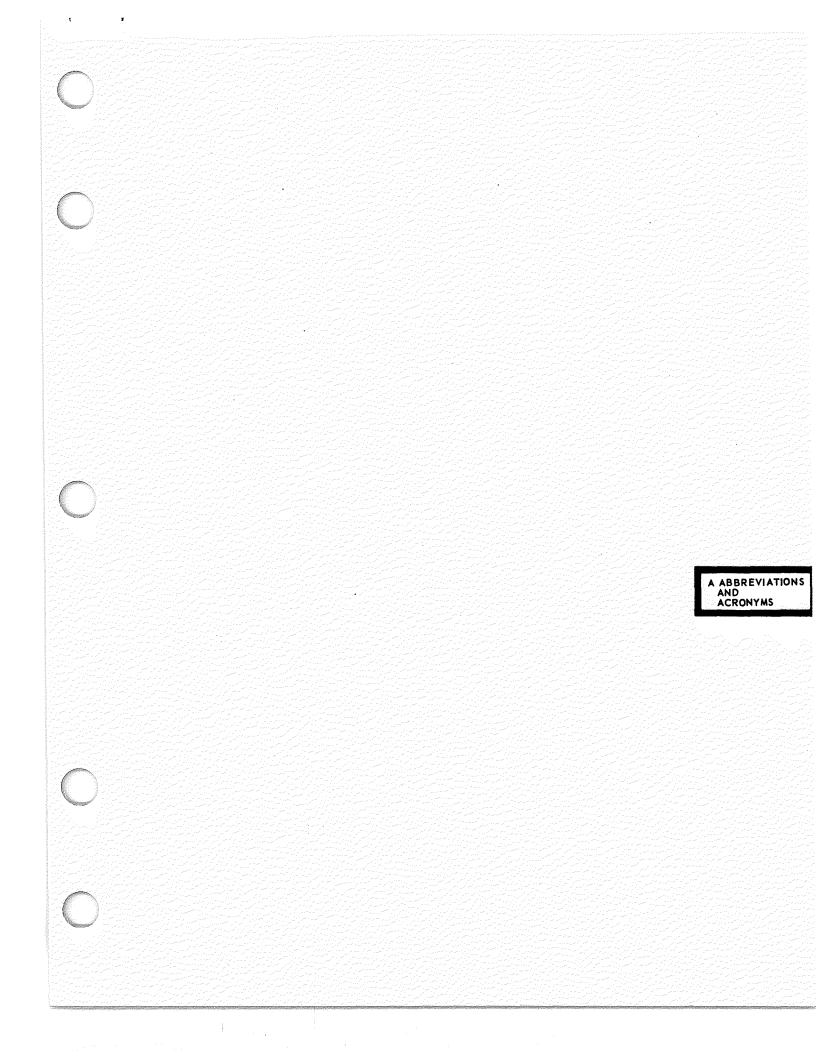
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MISSION RULES

SECTION 6 -	P&FS	SPECIFIC	RULES -	CONCLUDED
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R	RULE	CONDITION/MALFUNCTION	PHASE		RULING	CONCLUDE	CUES/NOTES/COMMENTS
	33-251	EXPERIMENT CALIBRATION COMMAND FAILS IN ON STATE	LO	CONT	INUE MISSION		SOME DATA LOSS PRIMARILY IN MAGNETOMETER AND TELESCOPES
	33-252	MAGNETOMETER 2.5 V REFERENCE ERROR	LO	CONT	INUE MISSION		MAGNETOMETER DATA WILL REQUIRE CORRECTION.
	33-253	,	LO	CONT	INUE MISSION		DATA MAY REQUIRE CORRECTION.
	33-254	LOSS OF MAGNETIC ZERO CROSSING	Γ̈́Ο	CONT	INUE MISSION		THE MAGNETIC PHASE COUNTER AND SECTOR GENERATORS WILL BE INOPERATIVE, INHIBITING DATA FROM EA A-4 WHILE IN TS MODES.
		MISSION APOLLO 16		DATE 12/15/71	SECTION SECURIC	GROUP	PAGE
		APOLLO 16	I'm	12/15//1	P&FS SPECIFIC RULES	MAGNETON	METER 6-6 Tape 3-11

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APPENDIX A - ABBREVIATIONS AND ACRONYMS

	MISSI APOLL		DATE 12/15/71	SECTION ABBREVIATIONS AND ACRONYMS	GROUP	PAGE A-1	
	Mico	ON DEW	DATE	CECTION	CDOUD	DACE 1	
	555	DSS/A DSS/D DSS/PROC	ANALOG DIGITA	COMPONENTS INCLUDE: 5 DATA PROCESSOR AL DATA PROCESSOR ETE DATA PROCESSOR (REDUNDANT)		
	D/P DPLY DRT DSS	DATA PRO DEPLOY DOME REM	OVAL TOOL	COMPONENTS INCLUDE:			
	DIG DIR/V DISSIP DLAY	DIGITAL DIRECTION DISSIPATE DELAY		ED (USED ON PSE)			
	DB DBM DC DEC DET	DIRECT (DECODER DETECTOR	WITH RESE CURRENT	PECT TO ONE MILLIWAT	Т		
	C.A.	COMMAND CLOCKWIS		ION WORD			
	CPS CS CTL CVR	CYCLES I CENTRAL CONTROL COVER	PER SECOND STATION				
	CONV CPLEE CPE	CONVERTE CHARGED		EXPERIMENT (FULL NAM	E IS CHARGED-P	ARTICLE LUNAR E	NVIRONMENT EXPERIMENT)
	CMD CNT CNTR COMM	CHAN/2 CHAN/HI CHAN/LO COMMAND COUNT COUNTER COMMUNIC	CHANNEI CHANNEI CHANNEI	TRON P/S NO. 2 TRON VOLTAGE INCREA TRON VOLTAGE INCREA	SES OFF SES OFF		
	CCIG COW CH CHAN	COUNTERC CHANNEL CHANNEL CHAN/1	LOCKWISE RON - USEI	GAGE (INSTRUMENT POR D IN CPE AS: LTRON P/S NO. 1	TION OF CCGE)		
	CCGE/A CCGE/D	ANALOG /	ND DIGITAL	ID READOUT FROM CC	GE	T AND 4, SEPARA	ATE MSC EXPERIMENT ON ALSE
	CAL CB CCGE	CALIBRAT CIRCUIT COLD CAT	BREAKER	FYDEDIMENT /DADT OF	SIDE ON ALCED	3 AND A CEDAR	ATE MCC EVDEDYNESS ON THE
	BCC BL BAS BER BPS		R RATE	NTROL F STRUCTURE TEMPERAT	URE		
	ASE AUTO AZ		SEISMIC EXF	PERIMENT			
	ALIGN ALSEP A/F AMPS ANT			ACE EXPERIMENTS PACK	AGE		
	A/DC ADC ADD ALCS	AMPERES ADDRESS ALSEP CO	DC OMPUTER SYS	CONVERTER			
	AC	ALTERNAT	ING CURREN	NT			
			ABBRE	VIATIONS AND ACRONYM	<u> </u>		
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MISSION RULES

APPENDIX A - ABBREVIATIONS AND ACRONYMS - CONTINUED

_ T		APPEND	IX A - ABBKE	VIATIONS AND ACRONYM	S - CONTINUED		
R ITEM							
	EA	FLECTR	OSTATIC ANAL	/7FD			
	EPS		ICAL POWER S				
1 1	EVA EXP	EXTRAV EXPERI	EHICULAR ACT	YTIVI			
	E.M.	LAFLINI	rica i				
	F	FAHREN	UCIT				
1 1	FET		EFFECT TRANS	ISTOR			
1 1	FLD	FIELD					
1 1	FMR FREQ	FREQUE	MISSION RUL NCY	: 2			-
1 1	FTT`		RANSFER TOOL				
	GDT	GRADIE	NT SENSOR DE	LTA TEMPERATURE (HFE)		
	GEO GLA	GEOPHO GRENAT	NE E LAUNCHER A	CCEMBIA			
1 1	GMBL	GIMBAL		SSCRIBET			
1 1	GND GRS	GROUND	RAY SPECTROM	TTED.			
1 1	GT	GRADIE	NT SENSOR AM	ETER BIENT TEMPERATURES (HFE)		
	HBR	HIGH B	IT RATE				
	HFE HTD	HEAT F	LOW EXPERIME	NT			
	HTR	HEATER HTR/HK	- UN HEE, T HIGH COND	HERE ARE TWO CASES: UCTIVITY HEATER			
	HT/S	HTR/LK	LOW CONDU	CTIVITY HEATER			
1 1	HT/S HV	HEAT S	INK OLTAGE				
	HVPS	HIGH V	OLTAGE POWER	SUPPLY			
	HZ	HERTZ					
1 1	****						
1 1	INST Insul	INSTRU INSULA					
	INT	INTERN					
1 1							
	JETTS	JETT IS	ON				
	K	KELVIN					
	KC KHZ	KILOCY					
	ΚV	KILOVO					
1 1							
	LAT LBR	LATITU					
1 1	LEO		T RATE EXPERIMENTS	OFFICER			
lll	LM	LUNAR	MODULE				
	LO L.O.	LUNAR	ORBIT OSCILLATOR				
	LONG	LONGIT	UDE				
	LOS LP	LOSS (F SIGNAL	ENCORC)			
	LRV	LUNAR	ERIOD (PSE S ROVING VEHIC	LE			
]	LSB LSD		SIGNIFICANT SIGNIFICANT				
	LSM		SURFACE MAGN				
] }	LVL	LEVEL					
	MA MALF	MILLIA					
1 1	MAP	MALFUN MESSAG	ICTION E ACCEPTABLE	PULSE			
	MC	MEGACY	CLE				
	MCC MDAC		IN CONTROL CE IPERES DC	NIER			
1 1	MDE	MODE					
	MEV MHZ	MILLIC MEGAHE	N ELECTRON V RT7	OLTS			
	MOCR	MISSIC	N OPERATIONS	CONTROL ROOM			
1 1	MOD MODE	MODULE OPERAT		E DEFINED AS FOLLOWS	FOR HEF.		
		MODE/0	GRADIEN	T MODE			
		MODE/L		NDUCTIVITY MODE DUCTIVITY MODE			
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MISSION RULES

APPENDIX A - ABBREVIATIONS AND ACRONYMS - CONTINUED

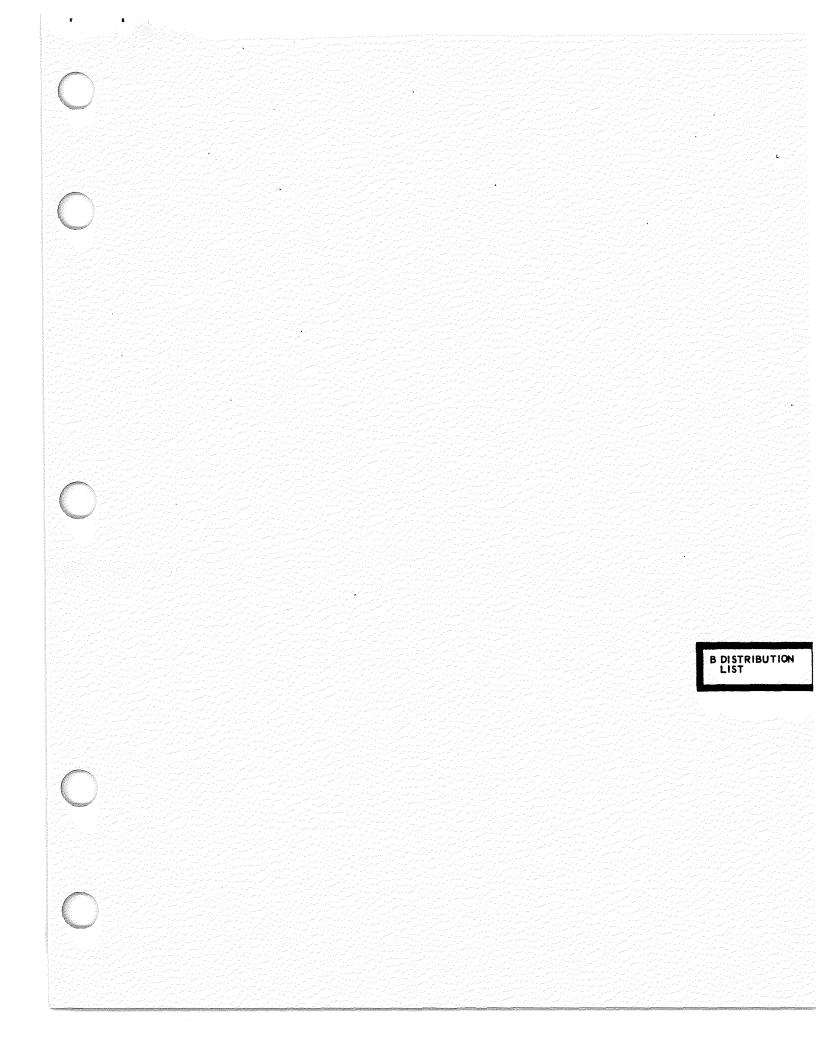
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K ITEM								
	MRO	MEMOR	RÝ RE.	ADOUT MODE	E			
	MS	MILLI	SECO	ND				
	MSB MSD			IFICANT B IFICANT D				
	MSFN	MANNE	D SP	ACE FLIGHT	T NETWORK			
	MTR MUX	MOTOF MULTI	0 - 0	N PSE, TH	E THREE MOTORS ARE I	MTRX, MTRY, AND	MTRX.	
	MV	MILL	VOLT	S				
	MW/CM2				ARE CENTIMETER			
	NA NBD	NANOA						
	NBR,	NURMA	AL BI	T RATE				
	OPER OSC	OPER/ OSCII		R				
	PA PCM			LIFIER; P E MODULAT	ICOAMPERES			
	PCU			DITIONING				
	PDR	POWER	RDIS	SIPATION	RESISTOR			
	PDU PET	POWE	R DIS AGE F	TRIBUTION LAPSED TI	UNIT MF			
	P&FS	PART:	ICLES	AND FIEL	D SUBSATELLITE			
	PHYS	PHYS: PHYS,			USED AS FOLLOWS:	VCCEMBIA)		
	PLT	PLAT	Ξ		L ANALYZER (SENSOR	MODEMBET)		
	PM PRE/LIM			ULATION				
	P/S	PRE - I POWE						
	PSE	PASS:	[VE S	EISMIC EX	PERIMENT - ALSO:			
		PSE/I PSE/:			PERIOD SENSORS -PERIOD SENSORS			
		PSE/I	P/SP	LONG-	AND SHORT-PERIOD S	ENSORS		
		LONG- THE	-PERI TWO H	OD SENSOR	S ARE FURTHER DEFIN LONG-PERIOD SENSOR	ED AS PSE/X, PS	SE/Y, AND PSE/Z,	WHILE PSE XY DENOTES
	R RCVR	RESI! RECE		(USED AS	R1 AND R2)			
	RDT	RING	SENS		TEMPERATURE (HFE)			
	REF RF	REFEI		QUENCY				
	R/S	REMO ⁻	TE SI					
	· RST RT	RESE*		OD AMRTEN	T TEMPERATURE (HFE)	. DEAL TYME		
	RTC	REAL-	-TIME	COMMAND		, REAL TIME		
	RTG RX	RADIO RECE		OPE THERM	OELECTRIC GENERATOR			
	SCI	SCIE	NTIFI	С				
	SEL SEQ	SELEC		SEQUENTI	۸۱۰			
	JLY	USED	ON H	FE ÀS:				
		SEQ/I SEQ/I	FUL	FULL SE	QUENCE SEQUENCE			
		SEQ/I	2	PROBE 2	SEQUENCE			
		USED	ON A	SE AS:	·			
	SEQ	SEQ/:	NTIFI	C EQUIPME	IAL SINGLE NT			
	SIDE	SUPRA	ATHER	MAL ION D	ETECTOR EXPERIMENT			
	SNSR SP	SENSI SHOR	T PER	IOD (PSE	SENSOR)			
	SPST	SING	_E PO	LE SINGLE	THROW			
	S/S SW	SAMPI SWIT		EK SECOND	; SIGNAL STRENGTH			
	SWS	SOLAI	RWIN	D SPECTRO	METER			
	SYNC	2146	1KUN L	ZATION				
l			REV	DATE	SECTION	GROUP	PAGE	
	AP0	LLO 16	FNL	12/15/71	ABBREVIATIONS		A-3	-
		1	- 1		AND ACRONYMS	I	1 1	Tape 4

MISSION RULES

APPENDIX A - ABBREVIATIONS AND ACRONYMS - CONCLUDED

		APPENDIX A - ABBF	EVIATIONS AND ACRONYMS	- CONCLUDED		
RITEM						-
	TC TEC TLC TM TS TSF TSN TX	THERMOCOUPLE (ON TRANSEARTH COAST TRANSLUNAR COAST TELEMETRY TELEMETRY STORAGE TELEMETRY STORE FOR TELEMETRY STORE FOR TRANSMITTER	AST	T TEMPERATURES	ARE READ ON EACH PR	OBE)
	USB UVP	UNIFIED S-BAND UNDERVOLTAGE PROT	ECTION CIRCUIT			
	V VCO V/FILT	VOLTAGE CONTROLLE	INDICATE "SPEED" ON P D OSCILLATOR A COMPONENT OF SIDE	SE IN "LVL DIR	/V")	
	W W1, W2, W3	WATTS WALL LOCATIONS OF	STRUCTURE TEMPERATURE	SENSORS .		
	XMTR XTAL XYZ XYO	TRANSMITTER CRYSTAL AXES OF LSM, WHER X, OR Y, OR NEITH	E XYO INDICATES ER			
1						
	MISSION APOLLO		SECTION ABBREVIATIONS	GROUP	PAGE A-3	

TSG 291A



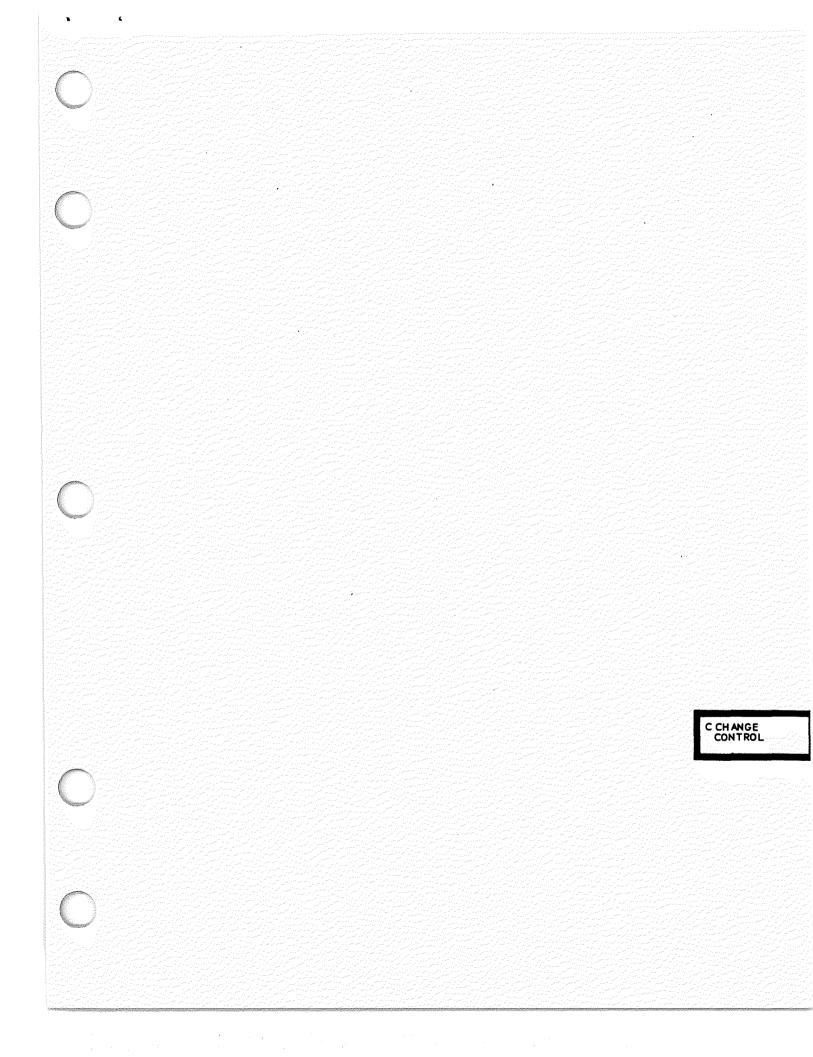
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APPENDIX B - DISTRIBUTION LIST

R ITEM		
3 es		
	DISTRIBUTION LIST	
	DEPUTY DIRECTOR	
A	AA/KRAFT, JR., C.C.	
	DIRECTOR OF FLIGHT OPERATIONS FA/SJOBERG, S. A.	
	를 받았다. 	
	FLIGHT CONTROL DIVISION FC/KRANZ, E. F.	
A	ROACH, J. W. FRANK, M. P.	
A A A	GRIFFIN, G. D. WINDLER, M. L. FC2/HARLAN, C. S. (6), PLUS SEND ALL EXTRAS TO HARLAN/FC2	
A A	FC6/H00VER, R. A. (3) FC9/SAULTZ, J. E. (30)	
	FC12/TECHNICAL SUPPORT GROUP (3)	
	FLIGHT SUPPORT DIVISION	
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	MISSION PLANNING AND ANALYSIS DIVISION FM/MAYER, J. P.	
	HUSS, C. R. FM4/WOLLENHAUPT, W. R.	
	DIRECTOR OF FLIGHT CREW OPERATIONS	
	CA/SLAYTON, D. K. CB/ASTRONAUT OFFICE	
	를 가는다. 	
	FLIGHT CREW SUPPORT DIVISION CA/NORTH, W. J.	
	CD/ALLEN, L. D. (2) CD3/RICHARD, L. G.	
	CG5/O'NEILL, J. W. CEK/THOMPSON, L. E. (4)	
	APOLLO SPACECRAFT PROGRAM OFFICE	
	PA/MCDIVITT, J. A., MORRIS, O.	
	KUBICKI, R. PA2/ASPO FILES	
	PA231/TASH, H. (3) PD4/SEVIER, J. (2) PD7/KOHRS, R. H. (2)	
	PD9/CRAIG, J. W. PE/(2)	
	PG/ COULTAS, G. A. PT/ARABIAN, D.	
	PT3/DATA LIBRARY (8)	
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	APOLLO 16 A 3/1/72 DISTRIBUTION B-1	

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R ITEM			ISTRIBUTION LIST - C			
o 8885-						
	SAFETY OFFICE					
	SA/FRENCH, J. C.					
	DIRECTOR OF SCIENCE AND	APPLICATIONS				
		475				
	TD/SCIENCE MISSION SUPPO TD5/BATES, J. R. (27) TDX/MILEY, R. R. (BDX)	(6)				
	OFFICE MANNED SPACEFLIG	HT.				
	MAO/LAND, E. W. (20)					
	MANAGEMENT SERVICES DIV	<u>ISION</u>				
	JM2/HILL, E. J. (3)					
	BOEING DATA MANAGEMENT					
	HA 04 (3)					
950 BASSIN						
	MISSION	REV DATE	SECTION	GROUP	PAGE	



MISSION RULES

APPENDIX C - CHANGE CONTROL

ITEM											
				CHANGE CONTROL							
	1.0	INTRODUCTION									
	1.0	INTRODUCTION									
·	1.1	WILL INSURE THE	PROPER COORDINA AKING THEM), AN	ATION OF CHANGES, PR	OVIDE A RECORD (OF PROPOSED C	ALSEP MISSION RULES. THI HANGES (INCLUDING THE LE UPDATES BETWEEN REVISIO				
	1.2	EFFECTIVITY JUNE 17, 1971									
	2.0	CHANGE PROCEDUR	ES		•						
	2.1		S ARE SOLICITED	FROM ANY INDIVIDUAL HIEF, FLIGHT CONTROL			LID INPUT, ALL CHANGES WI				
	2.1.1										
	2.2	APPROVAL COORDINATION THE ORIGINATOR OF THE CHANGE MAY OBTAIN PRELIMINARY CONCURRENCES. FCOB WILL OBTAIN FORMAL CONCURRENCES FROM THE CHIEF, LUNAR EARTH/EXPERIMENTS BRANCH AND THE CHIEF, FLIGHT CONTROL DIVISION.									
	2.2.1										
	2.2.3										
	2.3	PUBLICATION AND DISTRIBUTION OF INTERIM CHANGES INTERIM CHANGES WILL BE DISTRIBUTED BIA AN ABBREVIATED DISTRIBUTION LIST CONSISTING OF THE MISSION CONTROTEAM, PERTINENT NASA ORGANIZATIONS, AND THE APPROPRIATE CONTRACTOR(S).									
		MISSION	REV DATE	SECTION	GROUP	PAGE					

MISSION RULES

APPENDIX C - CHANGE CONTROL - CONCLUDED

3.3.2	REVISIONS WILL						
3.3.2		:	AN "AS REC	QUIRED" BASIS.	•		
	DISTRIBUTION REVISIONS WILL	BE PRINTED	AND DISTRI	IBUTED ACCORDI	NG TO APPENDIX	В.	

